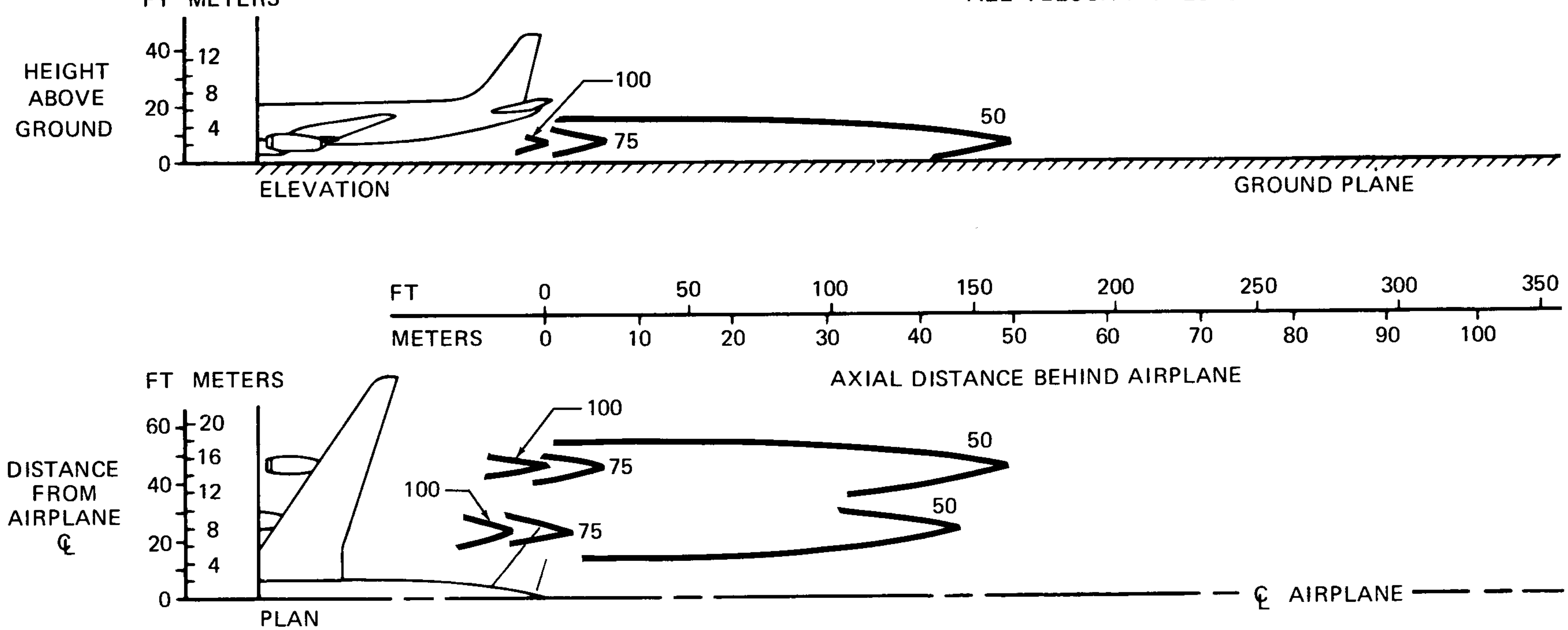
NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, JT3D, OR RC_O ENGINES
AVERAGE THRUST – JT3D @ 3400 LBS (1542 KG)
RC_O 12 – 3150 LBS (1429 KG)
ALL VELOCITY VALUES ARE MILES PER HR



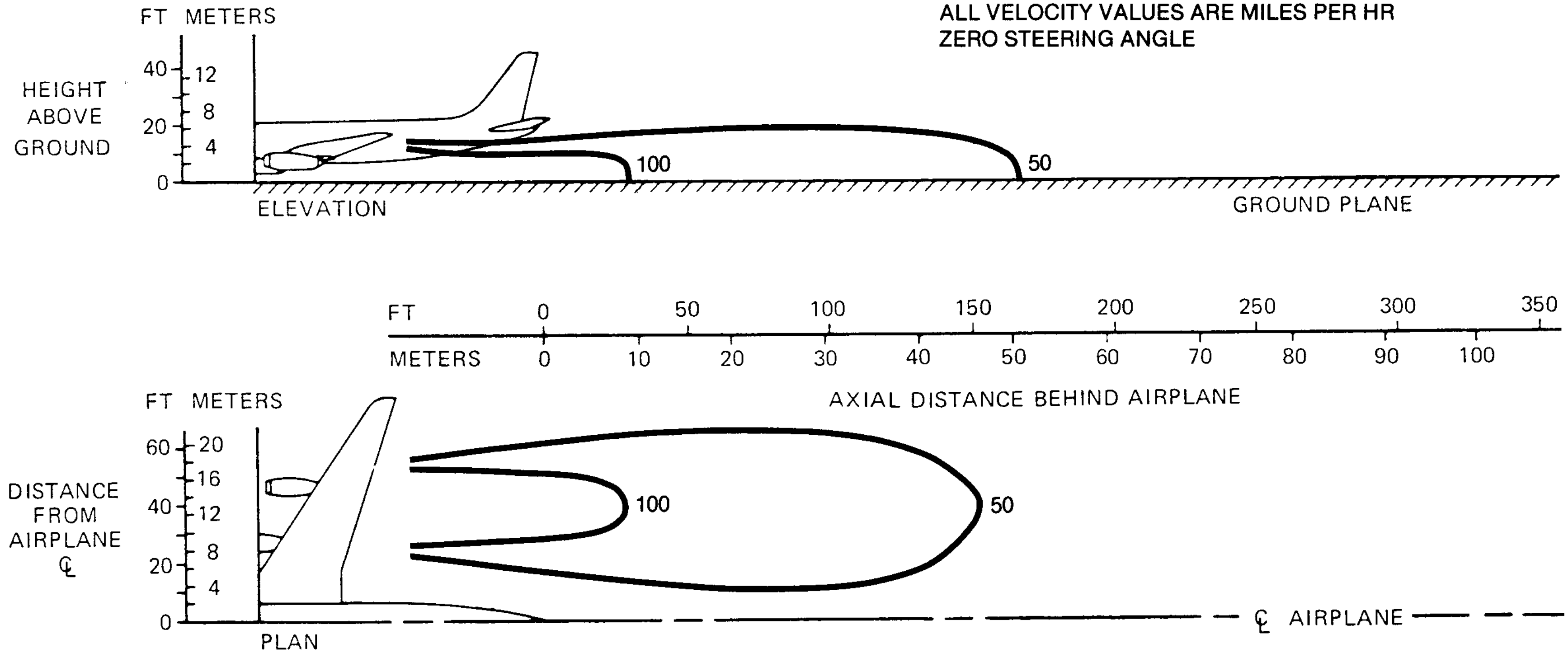
CONVERSION FACTOR: 1 MPH = 1.6 KM PER HR

6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.1 JET ENGINE EXHAUST VELOCITY CONTOURS, BREAKAWAY POWER MODEL DC-8, -43, -55, -61, -62, -63

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, CFM56-2C ENGINE
AVERAGE THRUST — 3,400 LB/ENGINE
THRUST SETTING TO OVERCOME COLD TIRE CONDITION
ALL VELOCITY VALUES ARE MILES PER HR
ZERO STEERING ANGLE



CONVERSION FACTOR: 1 MPH = 1.6 KM PER HR

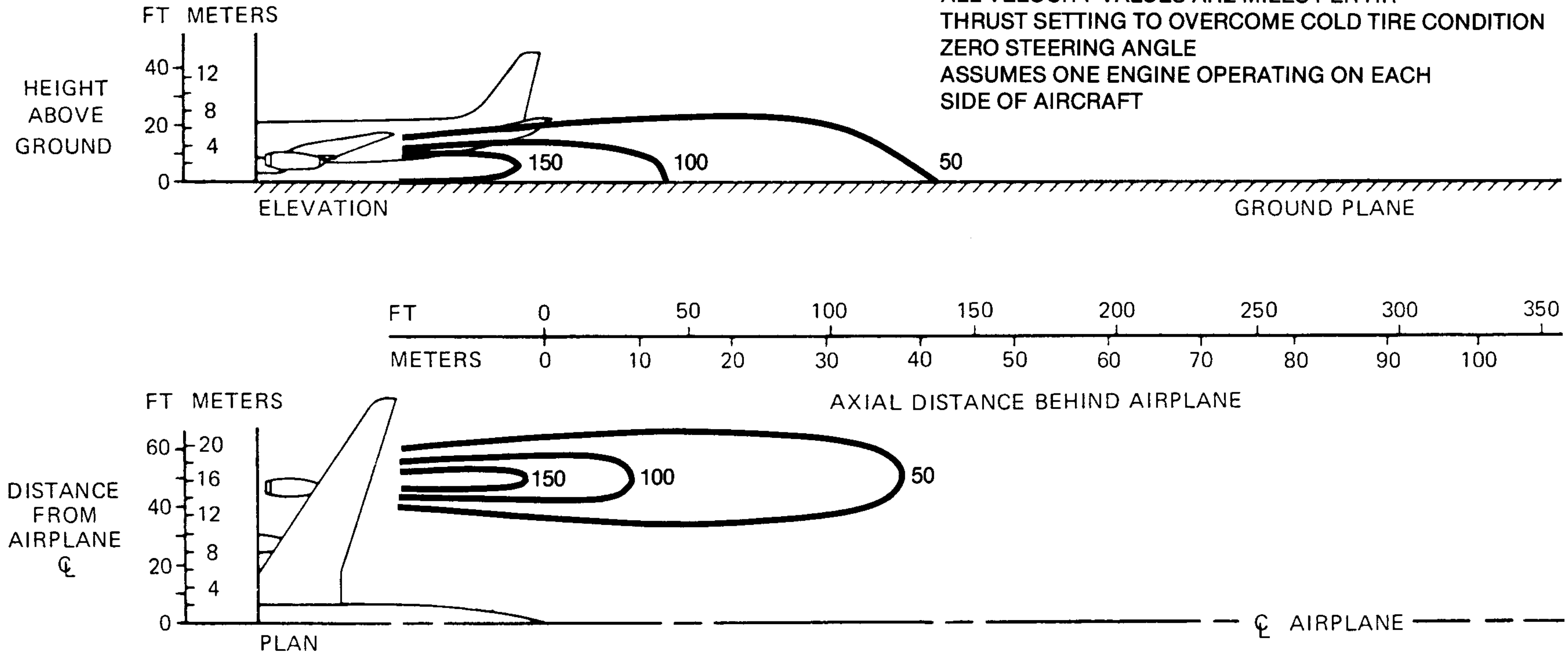
6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.1 JET ENGINE EXHAUST VELOCITY CONTOURS, 4 ENGINE BREAKAWAY POWER

MODEL DC-8, -71, -72, -73

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, CFM56-2C ENGINE
AVERAGE THRUST — 6,800 LB/ENGINE
ALL VELOCITY VALUES ARE MILES PER HR
THRUST SETTING TO OVERCOME COLD TIRE CONDITION
ZERO STEERING ANGLE
ASSUMES ONE ENGINE OPERATING ON EACH SIDE OF AIRCRAFT



CONVERSION FACTOR: 1 MPH = 1.6 KM PER HR

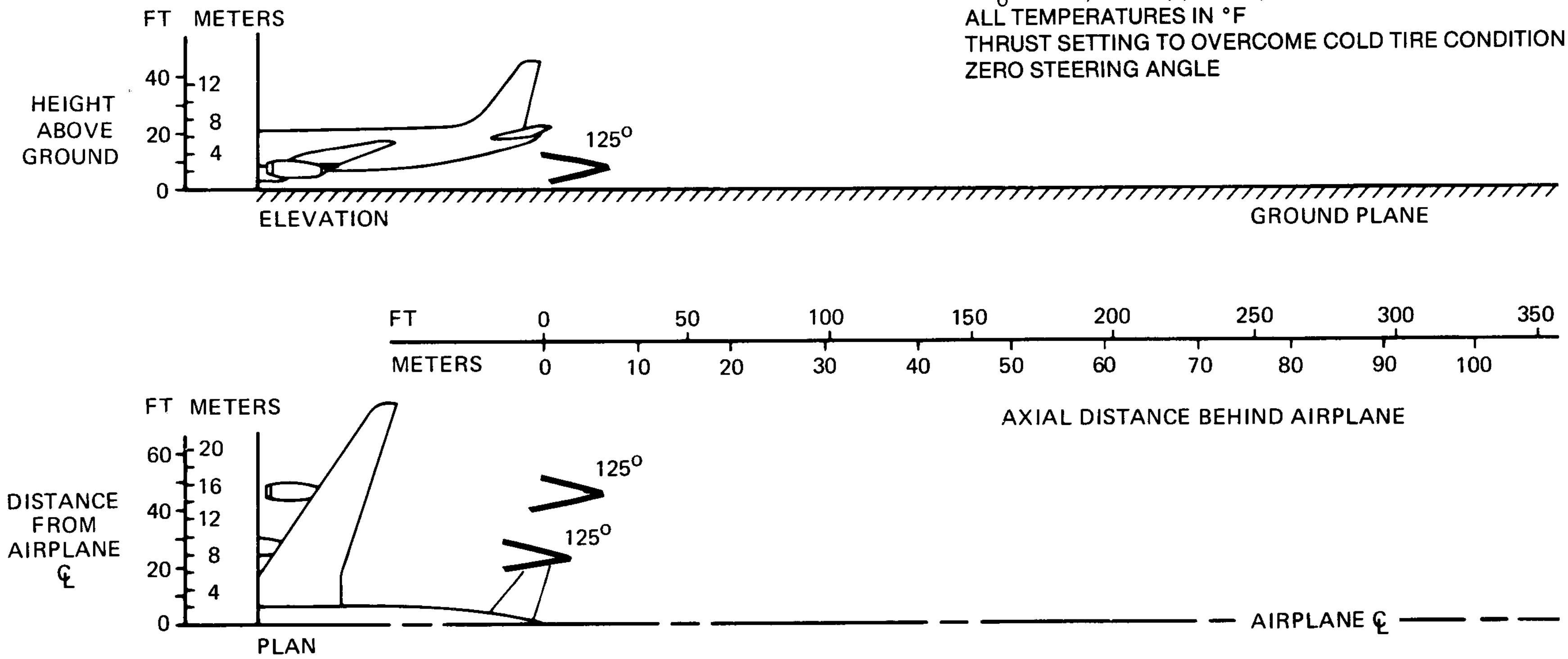
6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.1 JET ENGINE EXHAUST VELOCITY CONTOURS, 2-ENGINE BREAKAWAY POWER

MODEL DC-8, -71, -72, -73

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, JT3D OR RC_O 12 ENGINES
AVERAGE THRUST — JT3D — 3,400 LB (1,542 KG)
RC_O 12 — 3,150 LB (1,429 KG)
ALL TEMPERATURES IN °F
THRUST SETTING TO OVERCOME COLD TIRE CONDITION
ZERO STEERING ANGLE



CONVERSION: $°C = \frac{°F-32}{1.8}$

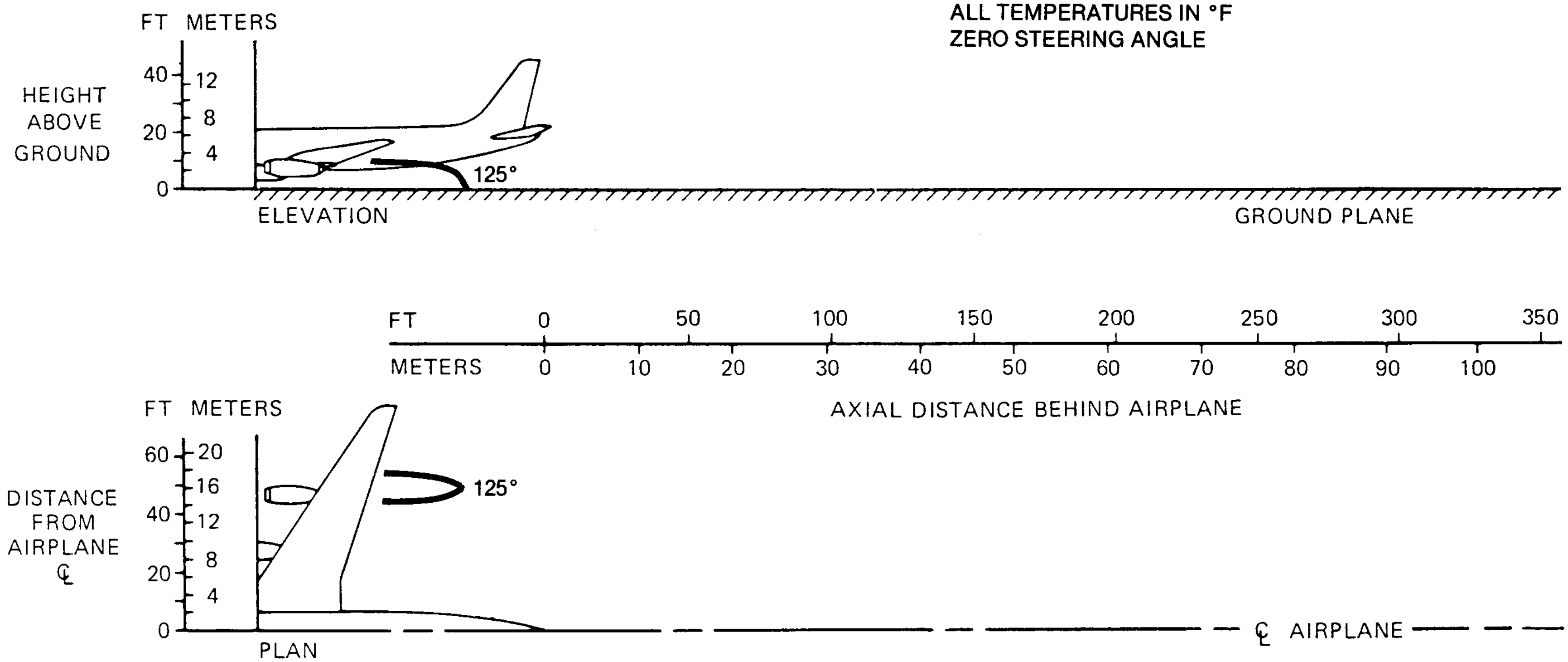
6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.2 JET ENGINE EXHAUST TEMPERATURE CONTOURS, BREAKAWAY POWER

MODEL DC-8, -43, -55, -61, -61F, -62, -62F, -63, -63F

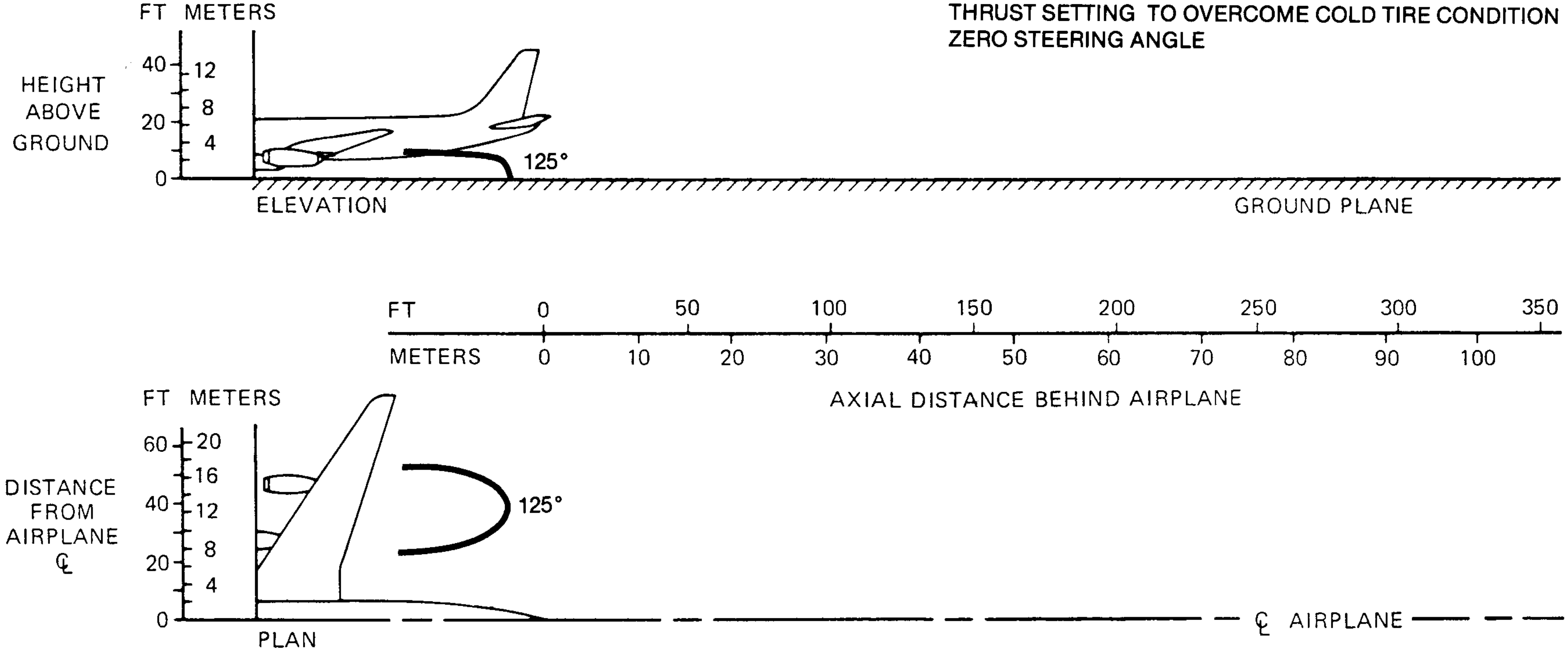
NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, CFM56-2C ENGINE
AVERAGE THRUST — 6,800 LB/ENGINE
THRUST SETTING TO OVERCOME COLD TIRE CONDITION
ALL TEMPERATURES IN °F
ZERO STEERING ANGLE



CONVERSION FACTOR: $°C = \frac{°F-32}{1.8}$

6.0 OPERATING CONDITIONS
6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES
6.1.2 JET ENGINE EXHAUST TEMPERATURE CONTOURS, 2-ENGINE BREAKAWAY POWER MODEL DC-8, -71, -72, -73

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, CFM56-2C ENGINE
ALL TEMPERATURES IN °F
AVERAGE THRUST — 3,400 LB/ENGINE
THRUST SETTING TO OVERCOME COLD TIRE CONDITION
ZERO STEERING ANGLE

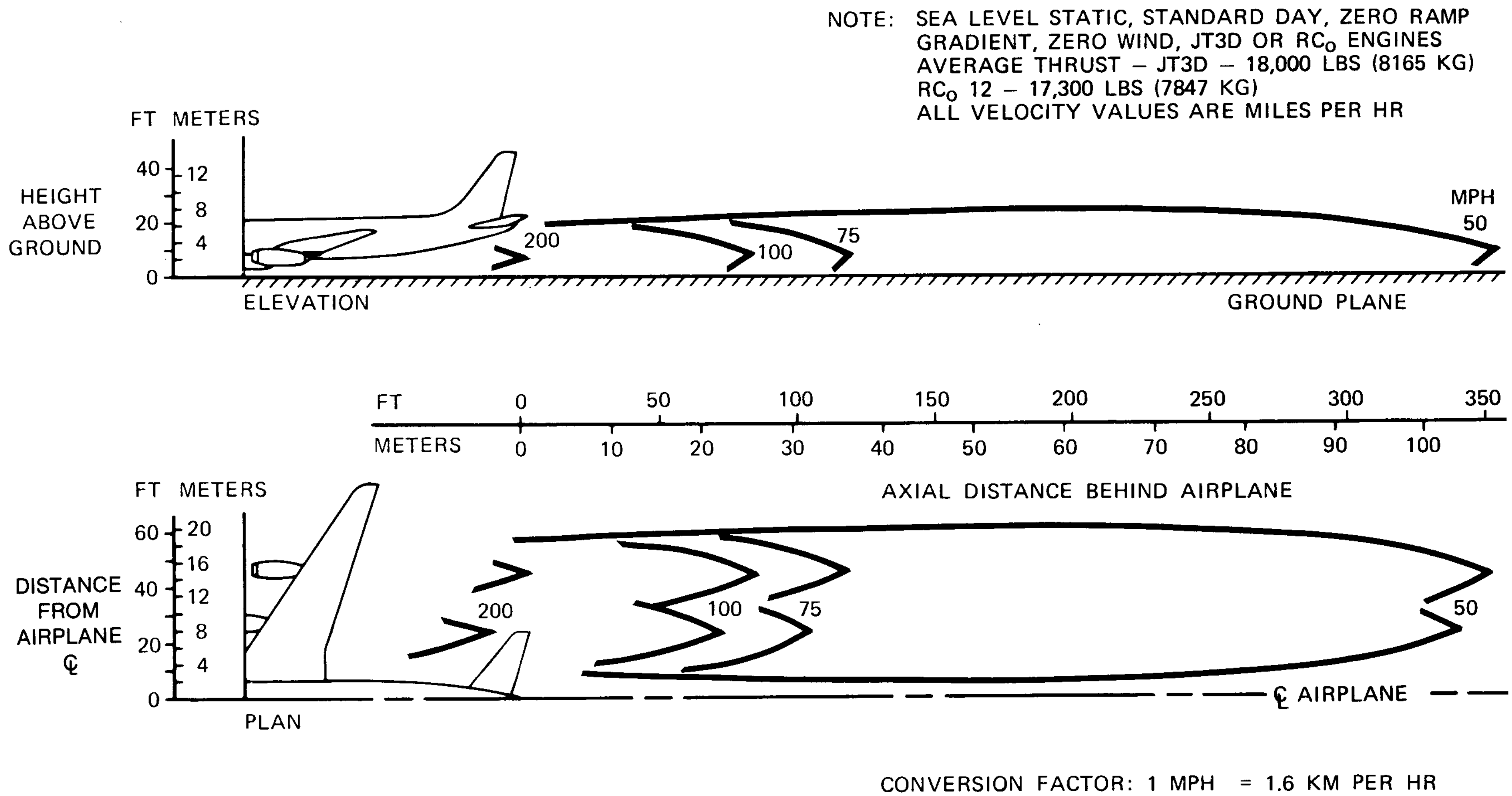


CONVERSION FACTOR: $°C = \frac{°F-32}{1.8}$

6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.2 JET ENGINE EXHAUST TEMPERATURE CONTOURS, 4-ENGINE BREAKAWAY POWER MODEL DC-8, -71, -72, -73



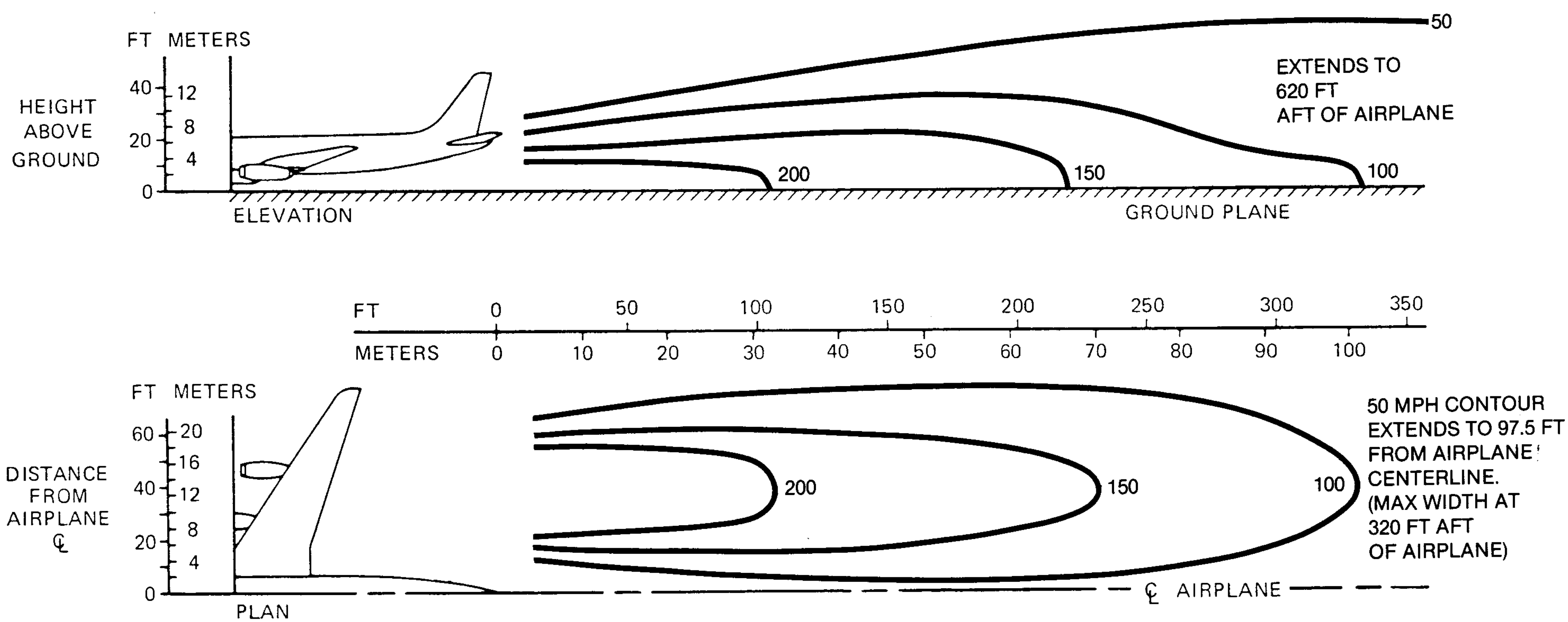
6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.3 JET ENGINE EXHAUST VELOCITY, TAKEOFF POWER

MODEL DC-8, -43, -55, -61, -61F, -62, -62F, -63, -63F

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, CFM56-2C ENGINE
AVERAGE THRUST — 22,000 LB/ENGINE
ALL VELOCITY VALUES ARE MILES PER HR

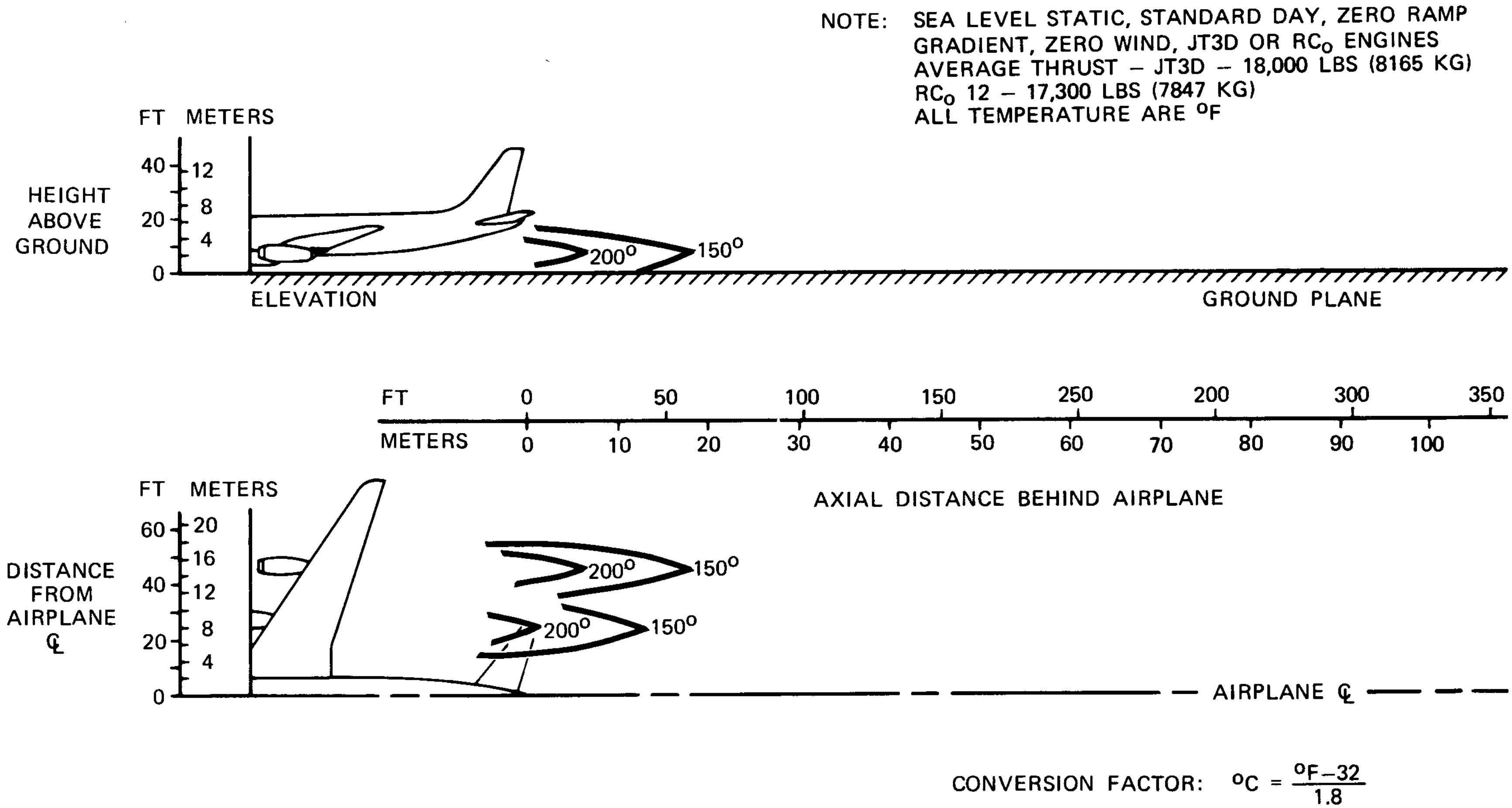


CONVERSION FACTOR: 1 MPH = 1.6 KM PER HR

6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.3 JET ENGINE VELOCITY CONTOURS, TAKEOFF POWER MODEL DC-8, -71, -72, -73

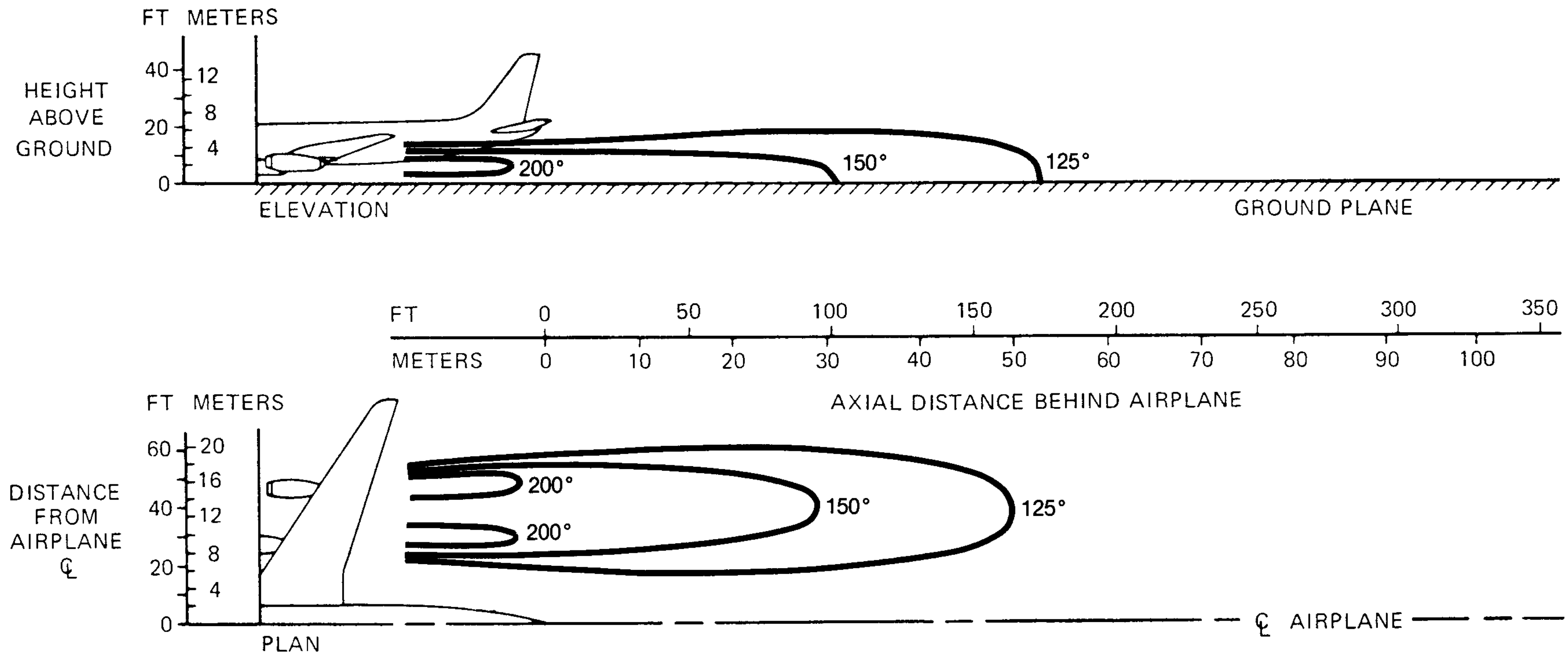


6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.4 JET ENGINE EXHAUST TEMPERATURE CONTOURS, TAKEOFF POWER MODEL DC-8, -43, -55, -61, -61F, -62, -62F, -63, -63F

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, CFM56-2C ENGINE
AVERAGE THRUST — 22,000 LB/ENGINE
ALL TEMPERATURES ARE °F



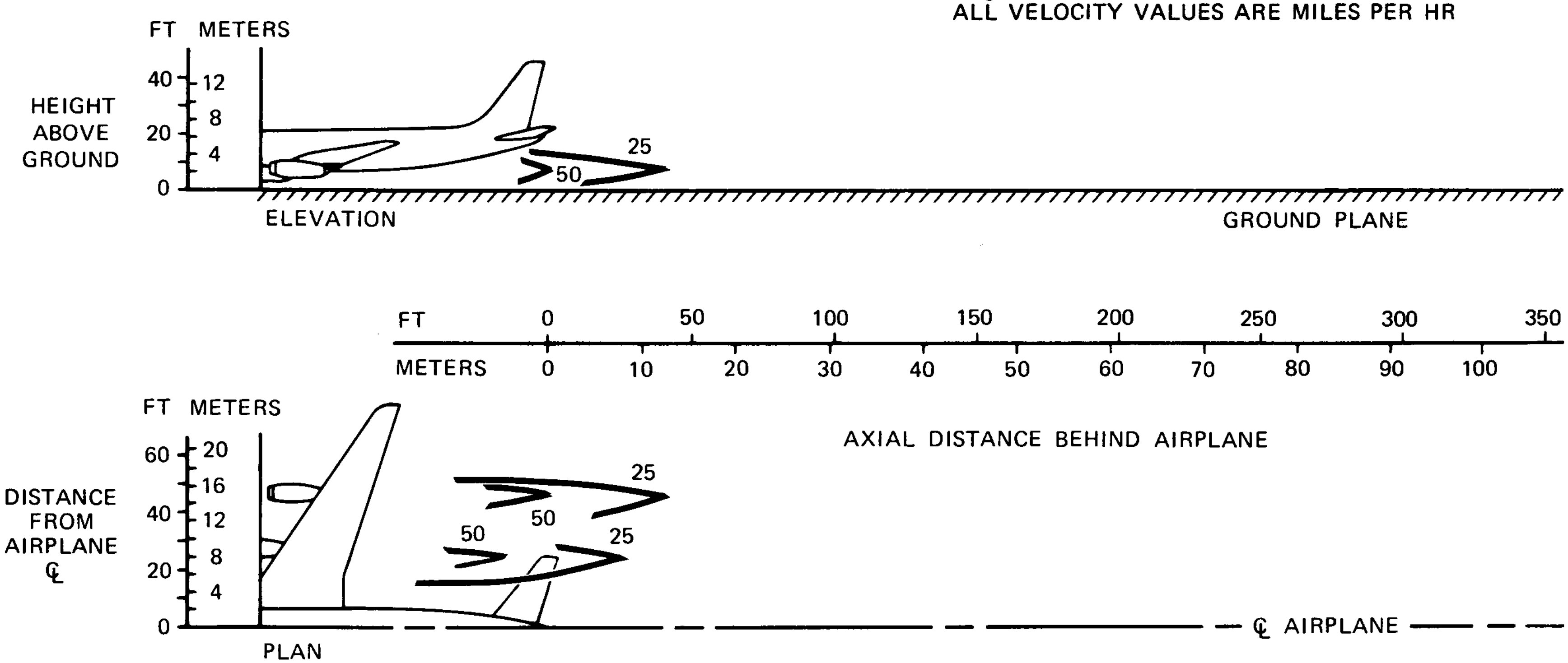
CONVERSION FACTOR: $°C = \frac{°F-32}{1.8}$

6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.4 JET ENGINE EXHAUST TEMPERATURE CONTOURS, TAKEOFF POWER MODEL DC-8, -71, -72, -73

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, JT3D OR RC_O ENGINES AVERAGE THRUST, JT3D – 900 LBS (408 KG) RC_O 12 – 760 LBS (345 KG) ALL VELOCITY VALUES ARE MILES PER HR



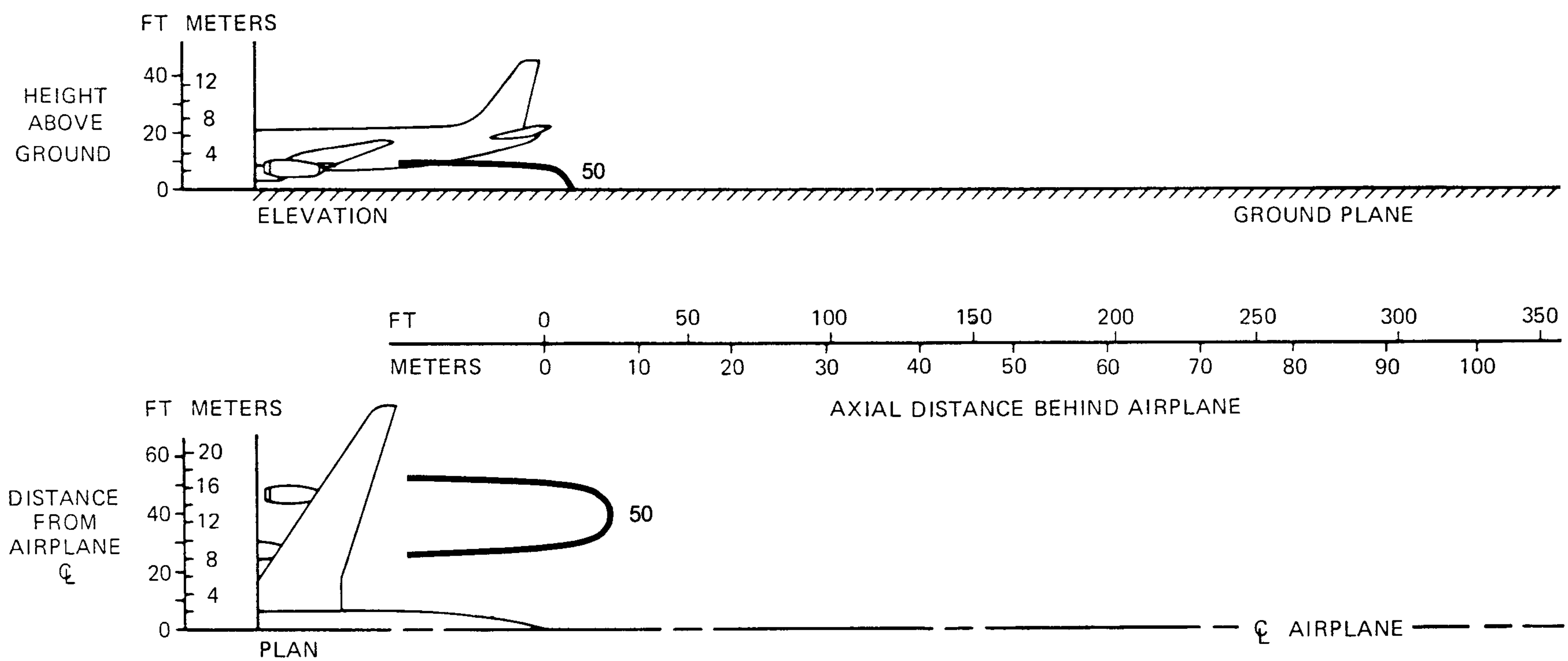
CONVERSION FACTOR: 1 MPH = 1.6 KM PER HR

6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.5 JET ENGINE EXHAUST VELOCITY CONTOURS, IDLE POWER MODEL DC-8, -43, -55, -61, -61F, -62, -62F, -63, -63F

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, CFM56-2C ENGINE
AVERAGE THRUST — 1,330 LB/ENGINE
ALL VELOCITY VALUES ARE IN MILES PER HR



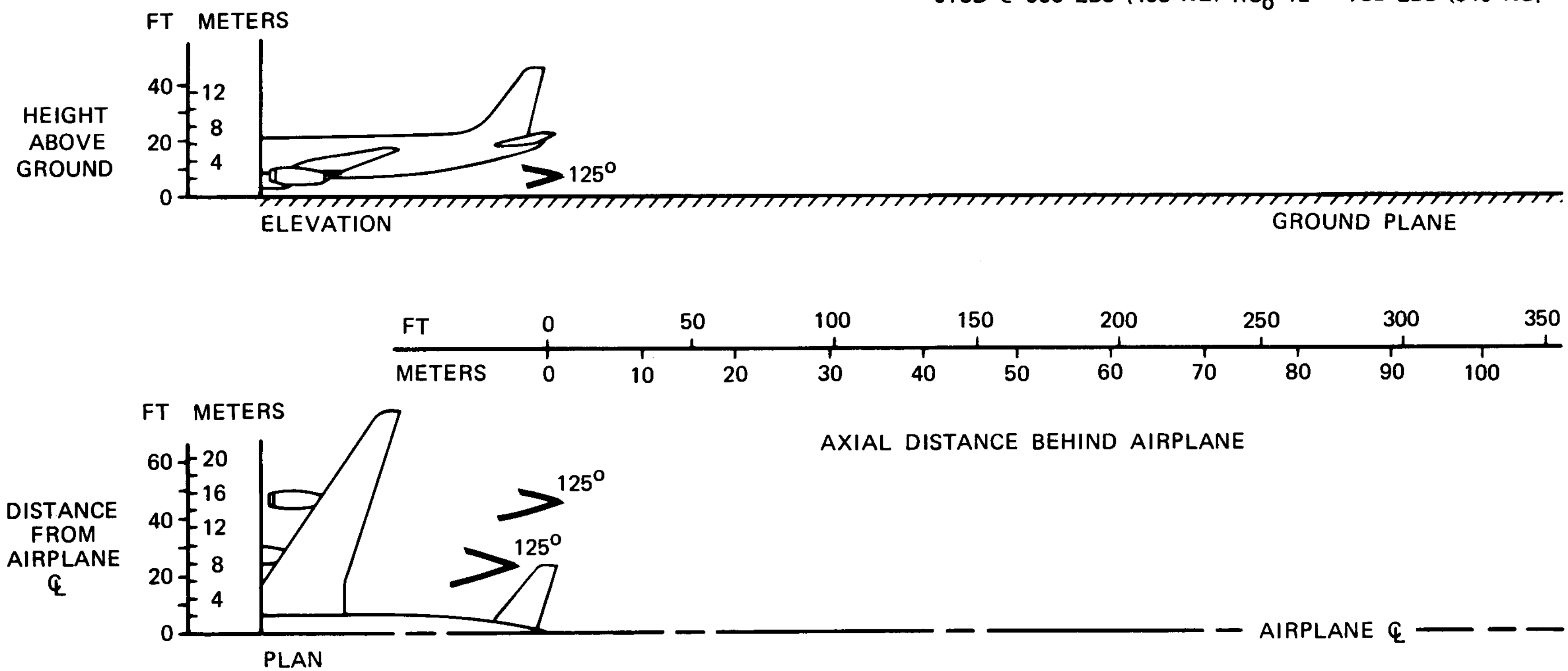
CONVERSION FACTOR: 1 MPH = 1.6 KM PER HR

6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.5 JET ENGINE EXHAUST VELOCITY CONTOURS, IDLE POWER MODEL DC-8, -71, -72, -73

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, JT3D OR RC_o ENGINES
ALL TEMPERATURES IN °F AVERAGE THRUST
JT3D @ 900 LBS (408 KG) RC_o 12 – 760 LBS (345 KG)



CONVERSION: $°C = \frac{°F - 32}{1.8}$

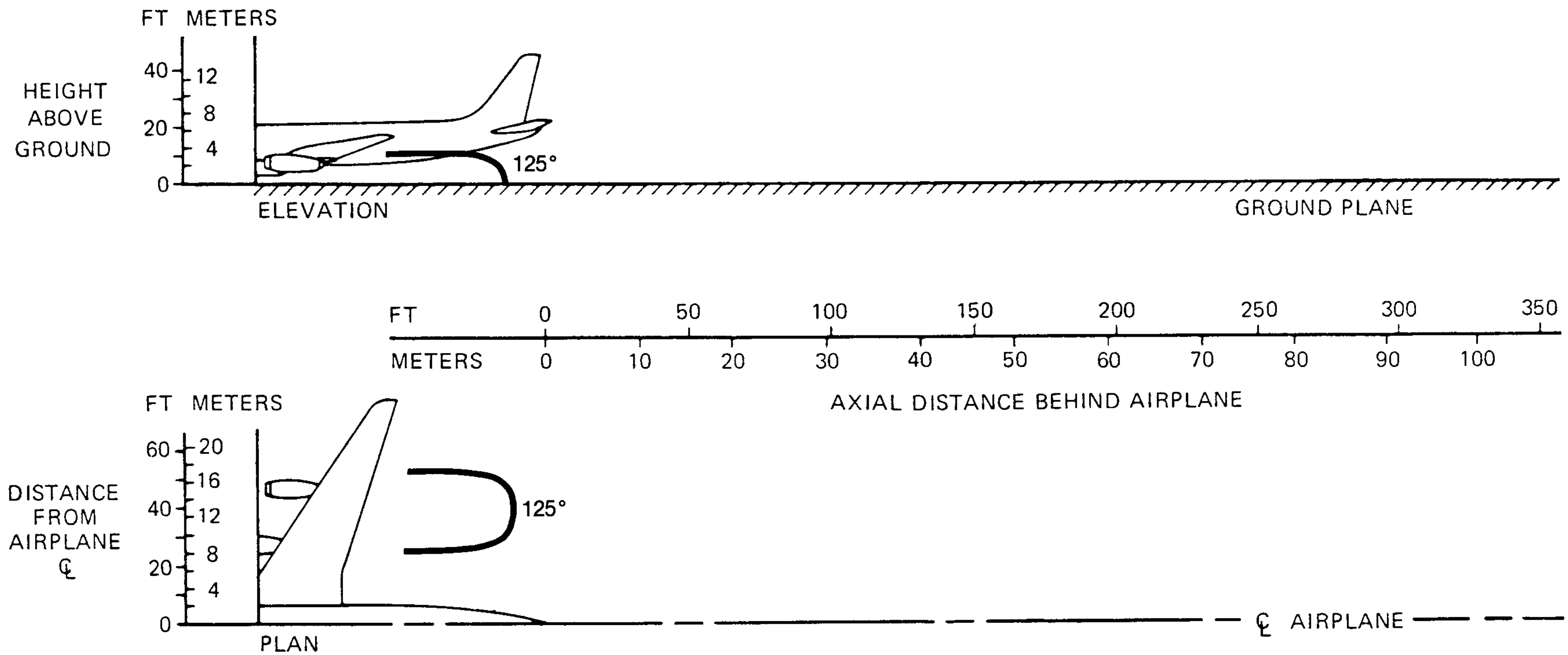
6.0 OPERATING CONDITIONS

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES

6.1.6 JET ENGINE EXHAUST TEMPERATURE CONTOURS, IDLE POWER

MODEL DC-8, -43, -55, -61, -61F, -62, -62F, -63, -63F

NOTE: SEA LEVEL STATIC, STANDARD DAY, ZERO RAMP GRADIENT, ZERO WIND, CFM56-2C ENGINE
ALL TEMPERATURES IN °F
AVERAGE THRUST — 1,330 LB/ENGINE



CONVERSION FACTOR: $°C = \frac{°F-32}{1.8}$

6.0 OPERATING CONDITIONS
6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURES
6.1.6 JET ENGINE EXHAUST TEMPERATURE CONTOURS, IDLE POWER
MODEL DC-8, -71, -72, -73

6.2 Airport and Community Noise

Aircraft noise is of major concern to the airport and community planner. The airport is a major element in the community's transportation system and, as such, is vital to its growth. However, the airport must also be a good neighbor, and this can be accomplished only with proper planning. Since aircraft noise extends beyond the boundaries of the airport, it is vital to consider the impact on surrounding communities. Many means have been devised to provide the planner with a tool to estimate the impact of airport operations. Too often they oversimplify noise to the point where the results become erroneous. Noise is not a simple subject; therefore, there are no simple answers.

The cumulative noise contour is an effective tool. However, care must be exercised to ensure that the contours, used correctly, estimate the noise resulting from aircraft operations conducted at an airport.

The size and shape of the single-event contours, which are inputs into the cumulative noise contours, are dependent upon numerous factors. They include:

1. Operational Factors

   a. *Aircraft Weight* — Aircraft weight is dependent on distance to be traveled, en route winds, payload, and anticipated aircraft delay upon reaching the destination.

   b. *Engine Power Settings* — The rates of ascent and descent and the noise levels emitted at the source are influenced by the power setting used.

   c. *Airport Altitude* — Higher airport altitudes will affect engine performance and thus can influence noise.

(Continued)

2. Atmospheric Conditions — Sound Propagation

   a. *Wind* — With stronger headwinds, the aircraft can take off and climb more rapidly relative to the ground. Also, winds can influence the distribution of noise in surrounding communities.

   b. *Temperature and Relative Humidity* — The absorption of noise in the atmosphere along the transmission path between the aircraft and the ground observer varies with both temperature and relative humidity.

3. Surface Condition — Shielding, Extra Ground Attenuation (EGA)

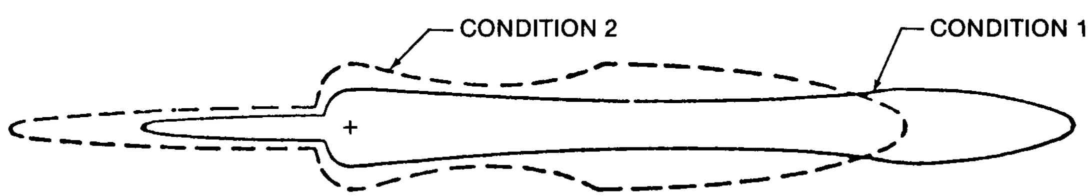
   a. *Terrain* — If the ground slopes down after takeoff or up before landing, noise will be reduced since the aircraft will be at a higher altitude above ground. Additionally, hills, shrubs, trees, and large buildings can act as sound buffers.

All of these factors can alter the shape and size of the contours appreciably. To demonstrate the effect of some of these factors, estimated noise level contours for two different operating conditions are shown below. These contours reflect a given noise level upon a ground level plane at runway elevation.

Condition 1

Landing:
- Maximum Design Landing Weight
- 10-knot Headwind
- 3° Approach
- 84°F
- Humidity 15%

Takeoff:
- Maximum Design Takeoff Weight
- Zero Wind
- 84°F
- Humidity 15%



Condition 2

Landing:
- 85% of Maximum Design Landing Weight
- 10-knot Headwind
- 3° Approach
- 59°F
- Humidity 70%

Takeoff:
- 80% of Maximum Design Takeoff Weight
- 10-knot Headwind
- 59°F
- Humidity 70%

As indicated from these data, the contour size varies substantially with operating and atmospheric conditions. Most aircraft operations are, of course, conducted at less than maximum gross weights because average flight distances are much shorter than maximum aircraft range capability and average load factors are less than 100 percent. Therefore, in developing cumulative contours for planning purposes, it is recommended that the airlines serving a particular city be contacted to provide operational information.

In addition, there are no universally accepted methods for developing aircraft noise contours or for relating the acceptability of specific noise zones to specific land uses. It is therefore expected that noise contour data for particular aircraft and the impact assessment methodology will be changing. To ensure that current available information of this type is used in any planning study, it is recommended that it be obtained directly from the Office of Environmental Quality in the Federal Aviation Administration in Washington, D.C.

It should be noted that the contours shown herein are only for illustrating the impact of operating and atmospheric conditions and do not represent the single-event contour of the family of aircraft described in this document. It is expected that the cumulative contours will be developed as required by planners using the data and methodology applicable to their specific study.