787 Airplane Characteristics for Airport Planning

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<tr>
<th>Revision Letter</th>
<th>Date</th>
<th>Changes in This Revision</th>
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<td>Table of Contents</td>
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<td>May 2011</td>
<td>Pages 3, 55</td>
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<td>December 2010</td>
<td>Page 100</td>
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<tr>
<td>C</td>
<td>July 2010</td>
<td>Pages 52-104</td>
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<td>B</td>
<td>December 2009</td>
<td>Pages i-ii, 7, 21-36, 75-98</td>
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</table>
# Table of Contents

1.0 SCOPE AND INTRODUCTION ........................................................................... 1-1
  1.1 SCOPE .............................................................................................................. 1-1
  1.2 INTRODUCTION ............................................................................................ 1-2
  1.3 A BRIEF DESCRIPTION OF THE 787 FAMILY OF AIRPLANES .......... 1-3
  1.4 CONVERSION FACTORS .............................................................................. 1-4

2.0 AIRPLANE DESCRIPTION.................................................................................. 2-1
  2.1 GENERAL CHARACTERISTICS ................................................................. 2-1
    2.1.1 General Characteristics: Model 787-8 ................................................... 2-2
    2.1.2 General Characteristics: Model 787-9 ................................................... 2-3
    2.1.3 General Characteristics: Model 787-10 ................................................. 2-4
  2.2 GENERAL DIMENSIONS .............................................................................. 2-5
    2.2.1 General Dimensions: Model 787-8 ........................................................ 2-5
    2.2.2 General Dimensions: Model 787-9 ........................................................ 2-6
    2.2.3 General Dimensions: Model 787-10 ...................................................... 2-7
  2.3 GROUND CLEARANCES .............................................................................. 2-8
    2.3.1 Ground Clearances: Model 787-8 .......................................................... 2-8
    2.3.2 Ground Clearances: Model 787-9 .......................................................... 2-9
    2.3.3 Ground Clearances: Model 787-10 ...................................................... 2-10
  2.4 INTERIOR ARRANGEMENTS .................................................................... 2-11
    2.4.1 Interior Arrangements - Typical: Model 787-8, 787-9, 787-10 ........... 2-11
  2.5 CABIN CROSS SECTIONS ...................................................................... 2-12
    2.5.1 Cabin Cross-Sections: Model 787-8, 787-9, 787-10 ............................ 2-12
  2.6 LOWER CARGO COMPARTMENTS ......................................................... 2-13
    2.6.1 Lower Cargo Compartments: Model 787-8, Containers and Bulk Cargo .......................................................... 2-13
    2.6.2 Lower Cargo Compartments: Model 787-9, Containers and Bulk Cargo .......................................................... 2-14
    2.6.3 Lower Cargo Compartments: Model 787-10, Containers and Bulk Cargo .......................................................... 2-15
  2.7 DOOR CLEARANCES .................................................................................. 2-16
    2.7.1 Door Locations: Model 787-8, 787-9, 787-10, Passenger and Cargo Doors .......................................................... 2-16
    2.7.2 Door Clearances: Model 787-8, 787-9, 787-10, Main Deck Entry and Service Doors .......................................................... 2-17
    2.7.3 Door Clearances: Model 787-8, 787-9, 787-10, Lower Deck Cargo Door (Forward & Aft) .......................................................... 2-18
    2.7.4 Door Clearances: Model 787-8, 787-9, 787-10, Bulk Cargo Door ..... 2-19

3.0 AIRPLANE PERFORMANCE .............................................................................. 3-1
  3.1 GENERAL INFORMATION ........................................................................... 3-1
3.2 PAYLOAD/RANGE FOR LONG RANGE CRUISE ..................................... 3-2

3.2.1 Payload/Range for Long-Range Cruise: Model 787-8 (Typical Engines) ................................................................................... 3-2

3.2.2 Payload/Range for Long-Range Cruise: Model 787-9 (Typical Engines) ................................................................................... 3-3

3.2.3 Payload/Range for Long-Range Cruise: Model 787-10 (Typical Engines) ................................................................................... 3-4

3.3 FAA/EASA TAKEOFF RUNWAY LENGTH REQUIREMENTS........... 3-5

3.3.1 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-8 (Typical Engines) .................................. 3-5

3.3.2 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 27°F (STD + 15°C), Dry Runway: Model 787-8 (Typical Engines) ................................................................................... 3-6

3.3.3 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-8 (Typical Engines) ................................................................................... 3-7

3.3.4 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 61°F (STD + 34°C), Dry Runway: Model 787-8 (Typical Engines) ................................................................................... 3-8

3.3.5 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-8 (Hi-Thrust Engines) .................. 3-9

3.3.6 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 27°F (STD + 15°C), Dry Runway: Model 787-8 (Hi-Thrust Engines) ................................................................................... 3-10

3.3.7 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-8 (Hi-Thrust Engines) ................................................................................... 3-11

3.3.8 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 61°F (STD + 34°C), Dry Runway: Model 787-8 (Hi-Thrust Engines) ................................................................................... 3-12

3.3.9 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-9 (Typical Engines) .................. 3-13

3.3.10 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 27°F (STD + 15°C), Dry Runway: Model 787-9 (Typical Engines) ................................................................................... 3-14

3.3.11 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-9 (Typical Engines) ................................................................................... 3-15

3.3.12 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 61°F (STD + 34°C), Dry Runway: Model 787-9 (Typical Engines) ................................................................................... 3-16

3.3.13 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-9, (Hi-Thrust Engines) ................. 3-17
3.3.14 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 27°F (STD + 15°C), Dry Runway: Model 787-9, (Hi-Thrust Engines) ................................................................. 3-18

3.3.15 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-9, (Hi-Thrust Engines) ................................................................. 3-19

3.3.16 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 61°F (STD + 34°C), Dry Runway: Model 787-9, (Hi-Thrust Engines) ................................................................. 3-20

3.3.17 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-10 (Typical Engines) ................. 3-21

3.3.18 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 27°F (STD + 15°C), Dry Runway: Model 787-10 (Typical Engines) ................................................................. 3-22

3.3.19 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-10 (Typical Engines) ................................................................. 3-23

3.3.20 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 61°F (STD + 34°C), Dry Runway: Model 787-10 (Typical Engines) ................................................................. 3-24

3.3.21 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-10 (Hi-Thrust Engines) ................. 3-25

3.3.22 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 27°F (STD + 15°C), Dry Runway: Model 787-10 (Hi-Thrust Engines) ................................................................. 3-26

3.3.23 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-10 (Hi-Thrust Engines) ................................................................. 3-27

3.3.24 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 61°F (STD + 34°C), Dry Runway: Model 787-10 (Hi-Thrust Engines) ................................................................. 3-28

3.4 FAA/EASA LANDING RUNWAY LENGTH REQUIREMENTS ............. 3-29

3.4.1 FAA/EASA Landing Runway Length Requirements – Flaps 30: Model 787-8 (All Engines) ................................................................. 3-29

3.4.2 FAA/EASA Landing Runway Length Requirements – Flaps 25: Model 787-8 (All Engines) ................................................................. 3-30

3.4.3 FAA/EASA Landing Runway Length Requirements – Flaps 30: Model 787-9 (All Engines) ................................................................. 3-31

3.4.4 FAA/EASA Landing Runway Length Requirements – Flaps 25: Model 787-9 (All Engines) ................................................................. 3-32

3.4.5 FAA/EASA Landing Runway Length Requirements – Flaps 30: Model 787-10 (All Engines) ................................................................. 3-33

3.4.6 FAA/EASA Landing Runway Length Requirements – Flaps 25: Model 787-10 (All Engines) ................................................................. 3-34
4.0 GROUND MANEUVERING ................................................................. 4-1
4.1 GENERAL INFORMATION ............................................................ 4-1
4.2 TURNING RADIIs ................................................................. 4-2
  4.2.1 Turning Radii – No Slip Angle: Model 787-8 ......................... 4-2
  4.2.2 Turning Radii – No Slip Angle: Model 787-9 ......................... 4-3
  4.2.3 Turning Radii – No Slip Angle: Model 787-10 ....................... 4-4
4.3 CLEARANCE RADII: MODEL 787-8, 787-9, 787-10 ....................... 4-5
4.4 VISIBILITY FROM COCKPIT IN STATIC POSITION ..................... 4-6
  4.4.1 Visibility from Cockpit in Static Position: Model 787-8, 787-9, 787-10 4-6
4.5 RUNWAY AND TAXIWAY TURNPATHS ........................................ 4-7
  4.5.1 Runway and Taxiway Turnpaths - Runway-to-Taxiway, More Than 90 Degree Turn: Model 787-8 ................................. 4-7
  4.5.2 Runway and Taxiway Turnpaths - Runway-to-Taxiway, 90 Degree Turn: Model 787-8 ......................................................... 4-8
  4.5.3 Runway and Taxiway Turnpaths - Taxiway-to-Taxiway, 90 Degree Turn: Model 787-8 ......................................................... 4-9
  4.5.4 Runway and Taxiway Turnpaths - Runway-to-Taxiway, More Than 90 Degree Turn: Model 787-9 ........................................ 4-10
  4.5.5 Runway and Taxiway Turnpaths - Runway-to-Taxiway, 90 Degree Turn: Model 787-9 ......................................................... 4-11
  4.5.6 Runway and Taxiway Turnpaths - Taxiway-to-Taxiway, 90 Degree Turn: Model 787-9 ......................................................... 4-12
  4.5.7 Runway and Taxiway Turnpaths - Runway-to-Taxiway, More Than 90 Degree Turn: Model 787-10 ........................................ 4-13
  4.5.8 Runway and Taxiway Turnpaths - Runway-to-Taxiway, 90 Degree Turn: Model 787-10 ......................................................... 4-14
  4.5.9 Runway and Taxiway Turnpaths - Taxiway-to-Taxiway, 90 Degree Turn: Model 787-10 ......................................................... 4-15
4.6 RUNWAY HOLDING BAY: MODEL 787-8, 787-9, 787-10 .......... 4-16

5.0 TERMINAL SERVICING ............................................................... 5-1
5.1 AIRPLANE SERVICING ARRANGEMENT - TYPICAL TURNAROUND .......................................................... 5-2
  5.1.1 Airplane Servicing Arrangement - Typical Turnaround: Model 787-8 ................................................................. 5-2
  5.1.2 Airplane Servicing Arrangement - Typical Turnaround: Model 787-9 ................................................................. 5-3
  5.1.3 Airplane Servicing Arrangement - Typical Turnaround: Model 787-10 ................................................................. 5-4
  5.1.4 Airplane Servicing Arrangement - Typical En Route: Model 787-8 ................................................................. 5-5
5.1.5 Airplane Servicing Arrangement - Typical En Route: Model 787-9 ................................................................. 5-6
5.1.6 Airplane Servicing Arrangement - Typical En Route: Model 787-10 ................................................................. 5-7
5.2 TERMINAL OPERATIONS - TURNAROUND STATION ................................................................. 5-8
5.2.1 Terminal Operations, Turntime Analysis - Turnaround Station: Model 787-8 ............................................................... 5-8
5.2.2 Terminal Operations, Turntime Analysis - Turnaround Station: Model 787-9 ............................................................... 5-9
5.2.3 Terminal Operations, Turntime Analysis - Turnaround Station: Model 787-10 ............................................................... 5-10
5.3 TERMINAL OPERATIONS - EN ROUTE STATION ................................................................. 5-11
5.3.1 Terminal Operations, Turntime Analysis - En Route Station: Model 787-8 ............................................................... 5-11
5.3.2 Terminal Operations, Turntime Analysis - En Route Station: Model 787-9 ............................................................... 5-12
5.3.3 Terminal Operations, Turntime Analysis - En Route Station: Model 787-10 ............................................................... 5-13
5.4 GROUND SERVICE CONNECTIONS ................................................................. 5-14
5.4.1 Ground Service Connections: Model 787-8 ................................................................................................. 5-14
5.4.2 Ground Service Connections and Capacities: Model 787-8 ........................................................................ 5-15
5.4.3 Ground Servicing Connections: Model 787-9 ................................................................................................. 5-16
5.4.4 Ground Servicing Connections and Capacities: Model 787-9 ........................................................................ 5-17
5.4.5 Ground Servicing Connections: Model 787-10 ................................................................................................. 5-18
5.4.6 Ground Servicing Connections and Capacities: Model 787-10 ........................................................................ 5-19
5.5 ENGINE STARTING AND GROUND POWER REQUIREMENTS ............................................ 5-20
5.5.1 Engine Starting Ground Power Requirements – Electrical – APU: Model 787-8, 787-9, 787-10 ................................................................. 5-20
5.5.2 Engine Starting Ground Power Requirements – Electrical – APU Inoperative – Two/Three GPU: Model 787-8, 787-9, 787-10 ................................................................. 5-21
5.5.3 Engine Power Requirements – Pneumatic: Model 787-8, 787-9, 787-10 ................................................................. 5-22
5.6 CONDITIONED AIR REQUIREMENTS ................................................................. 5-23
5.6.1 Conditioned Air Flow Requirements – Cooling Time: Model 787-8 ................................................................. 5-23
5.6.2 Conditioned Air Flow Requirements – Cooling – Steady State (103 F Ambient Air): Model 787-8 ................................................................. 5-24
5.6.3 Conditioned Air Flow Requirements – Cooling – Steady State (80 F Ambient Air): Model 787-8 ................................................................. 5-25
5.6.4 Conditioned Air Flow Requirements – Heating Time: Model 787-8 ................................................................. 5-26
5.6.5 Conditioned Air Flow Requirements – Heating – Steady State: Model 787-8 ................................................................. 5-27
5.6.6 Conditioned Air Flow Requirements – Cooling Time: Model 787-9 ............................................................................................................ 5-28
5.6.7 Conditioned Air Flow Requirements – Cooling – Steady State (103 F Ambient Air): Model 787-9 ................................................................. 5-29
5.6.8 Conditioned Air Flow Requirements – Cooling – Steady State (80 F Ambient Air): Model 787-9 ................................................................. 5-30
5.6.9 Conditioned Air Flow Requirements – Heating Time: Model 787-9 ............................................................................................................ 5-31
5.6.10 Conditioned Air Flow Requirements – Heating – Steady State: Model 787-9 .................................................................................................... 5-32
5.6.11 Conditioned Air Flow Requirements – Cooling Time: Model 787-10 ........................................................................................................ 5-33
5.6.12 Conditioned Air Flow Requirements – Cooling – Steady State (103 F Ambient Air): Model 787-10 ................................................................. 5-34
5.6.13 Conditioned Air Flow Requirements – Cooling – Steady State (80 F Ambient Air): Model 787-10 ................................................................. 5-35
5.6.14 Conditioned Air Flow Requirements – Heating Time: Model 787-10 ........................................................................................................ 5-36
5.6.15 Conditioned Air Flow Requirements – Heating – Steady State: Model 787-10 .................................................................................................... 5-37
5.7 GROUND TOWING REQUIREMENTS ................................................................................................................................. 5-38
5.7.1 Ground Towing Requirements - English Units: Model 787-8, 787-9, 787-10 ........................................................................................................ 5-38
5.7.2 Ground Towing Requirements - Metric Units: Model 787-8, 787-9, 787-10 ........................................................................................................ 5-39
6.0 JET ENGINE WAKE AND NOISE DATA ................................................................................. 6-1
6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURE .................................. 6-1
6.1.1 Jet Engine Exhaust Velocity Contours – Idle Thrust / Both Engines: Model 787-8 ........................................................................................................ 6-2
6.1.2 Jet Engine Exhaust Velocity Contours - Breakaway Thrust / 0% Slope / Both Engines / MTW: Model 787-8 ................................................................. 6-3
6.1.3 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 1% Slope / Both Engines / MTW: Model 787-8 ................................................................. 6-4
6.1.4 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Single Engine / MTW: Model 787-8 ................................................................. 6-5
6.1.5 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Single Engine / MLW: Model 787-8 ................................................................. 6-6
6.1.6 Jet Engine Exhaust Velocity Contours – Takeoff Thrust / Both Engines: Model 787-8 ........................................................................................................ 6-7
6.1.7 Jet Engine Exhaust Temperature Contours – Idle/Breakaway Thrust: Model 787-8 ........................................................................................................ 6-8
6.1.8 Jet Engine Exhaust Temperature Contours – Takeoff Thrust / Both Engines: Model 787-8 ........................................................................................................ 6-9
6.1.9  Jet Engine Exhaust Velocity Contours – Idle Thrust: Model 787-9 .... 6-10
6.1.10 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Both Engines / MTW: Model 787-9 ................................. 6-11
6.1.11 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 1% Slope / Both Engines / MTW: Model 787-9 ................................. 6-12
6.1.13 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Single Engine / MLW: Model 787-9 ................................. 6-14
6.1.15 Jet Engine Exhaust Temperature Contours – Idle/Breakaway Thrust: Model 787-9 ................................................................. 6-16
6.1.16 Jet Engine Exhaust Temperature Contours – Takeoff Thrust / Both Engines: Model 787-9 .............................................................. 6-17
6.1.17 Jet Engine Exhaust Velocity Contours – Idle Thrust / Both Engines: Model 787-10 ................................................................. 6-18
6.1.18 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Both Engines: Model 787-10 .............................................................. 6-19
6.1.19 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 1% Slope / Both Engines / MTW: Model 787-10 ................................. 6-20
6.1.20 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Single Engine / MTW: Model 787-10 ................................. 6-21
6.1.21 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Single Engine / MLW: Model 787-10 ................................. 6-22
6.1.22 Jet Engine Exhaust Velocity Contours – Takeoff Thrust / Both Engines: Model 787-10 .............................................................. 6-23
6.1.23 Jet Engine Exhaust Temperature Contours – Idle/Breakaway Thrust: Model 787-10 .............................................................. 6-24
6.1.24 Jet Engine Exhaust Temperature Contours – Takeoff Thrust / Both Engines: Model 787-10 .............................................................. 6-25
6.1.25 Inlet Hazard Areas: All Models .................................................. 6-26
6.2  AIRPORT AND COMMUNITY NOISE ....................................................... 6-27
7.0  PAVEMENT DATA .............................................................................. 7-1
7.1  GENERAL INFORMATION ................................................................. 7-1
7.2  LANDING GEAR FOOTPRINT: MODEL 787-8, 787-9, 787-10 .......... 7-4
7.3  MAXIMUM PAVEMENT LOADS: MODEL 787-8, 787-9, 787-10 ......... 7-5
7.4  LANDING GEAR LOADING ON PAVEMENT ................................. 7-6
7.4.1  Landing Gear Loading on Pavement: Model 787-8 ........................ 7-6
7.4.2  Landing Gear Loading on Pavement: Model 787-9 ........................ 7-7
7.4.3  Landing Gear Loading on Pavement: Model 787-10 ...................... 7-8
7.5 FLEXIBLE PAVEMENT REQUIREMENTS - FAA DESIGN METHOD

7.6 FLEXIBLE PAVEMENT REQUIREMENTS - LCN CONVERSION

7.7 RIGID PAVEMENT REQUIREMENTS - PORTLAND CEMENT ASSOCIATION DESIGN METHOD

7.8 RIGID PAVEMENT REQUIREMENTS - LCN CONVERSION

7.9 RIGID PAVEMENT REQUIREMENTS - FAA DESIGN METHOD

7.10 ACN/PCN REPORTING SYSTEM - FLEXIBLE AND RIGID PAVEMENTS

7.11 ACR/PCR REPORTING SYSTEM - FLEXIBLE AND RIGID PAVEMENTS

8.0 FUTURE 787 DERIVATIVE AIRPLANES

9.0 SCALED 787 DRAWINGS

9.1 MODEL 787-8

9.2 MODEL 787-9

9.3 MODEL 787-10

9.1.1 Scaled Drawings – 1:500: Model 787-8

9.1.2 Scaled Drawings – 1:500: Model 787-8

9.2.1 Scaled Drawing – 1:500: Model 787-9

9.2.2 Scaled Drawing – 1:500: Model 787-9

9.3.1 Scaled Drawings – 1:500: Model 787-10

9.3.2 Scaled Drawings – 1:500: Model 787-10
1.0 SCOPE AND INTRODUCTION

1.1 SCOPE

This document provides, in a standardized format, airplane characteristics data for general airport planning. Since operational practices vary among airlines, specific data should be coordinated with the using airlines prior to facility design. Boeing Commercial Airplanes should be contacted for any additional information required.

Content of the document reflects the results of a coordinated effort by representatives from the following organizations:

- Aerospace Industries Association
- Airports Council International - North America
- Air Transport Association of America
- International Air Transport Association

The airport planner may also want to consider the information presented in the "Commercial Aircraft Design Characteristics - Trends and Growth Projections," for long range planning needs and can be accessed via the following website:

http://www.boeing.com/airports

The document is updated periodically and represents the coordinated efforts of the following organizations regarding future aircraft growth trends.

- International Coordinating Council of Aerospace Industries Associations
- Airports Council International - North American and World Organizations
- Air Transport Association of America
- International Air Transport Association
1.2 INTRODUCTION

This document conforms to NAS 3601. It provides characteristics of the Boeing 787 Dreamliner family of airplanes for airport planners and operators, airlines, architectural and engineering consultant organizations, and other interested industry agencies. Airplane changes and available options may alter model characteristics. The data presented herein reflects the 787 family. Data used is generic in scope and not customer-specific.

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1.3 **A BRIEF DESCRIPTION OF THE 787 FAMILY OF AIRPLANES**

The 787 Dreamliner is an efficient family of twin-engine airplanes with exceptional environmental performance and new passenger-pleasing features. An international team of aerospace companies builds the 787, led by Boeing at its Everett facility near Seattle, WA and in North Charleston, SC.

787 Family

The 787 family is designed for medium- to long-range flights. In a typical dual-class configuration, the 787-8 can carry 242 passengers, the 787-9 can carry 290 passengers, and the 787-10 will carry 330 passengers.

787 Engines

The 787 features new engines from General Electric and Rolls-Royce that represent nearly a two-generation jump in technology.

Cargo Handling

The lower lobe cargo compartments can accommodate a variety of containers and pallets now in use.

Ground Servicing

The 787 features a more-electric design and does not have a traditional pneumatic system. The traditional pneumatic starters on the engines are replaced with a pair of gearbox-mounted main-engine starter/generators. Cabin air conditioning and wing anti-ice systems are also electrically powered. The remaining pneumatic system is for engine nacelle anti-ice. The airplane has ground service connections compatible with existing ground service equipment, and no special equipment is necessary. In case of an inoperable APU, engine starts may be accomplished via the airplane's external ground electrical connections.
1.4 CONVERSION FACTORS

The data in this manual is provided in both English and Metric units. Unless otherwise stated, the conversions listed below are used throughout this manual.

<table>
<thead>
<tr>
<th>MULTIPLY</th>
<th>BY</th>
<th>TO OBTAIN</th>
</tr>
</thead>
<tbody>
<tr>
<td>Pounds</td>
<td>0.45359237</td>
<td>Kilograms</td>
</tr>
<tr>
<td>U.S. Gallons</td>
<td>3.78541180</td>
<td>Liters</td>
</tr>
<tr>
<td>Inches</td>
<td>2.54000000</td>
<td>Centimeters</td>
</tr>
<tr>
<td>Feet</td>
<td>0.30480000</td>
<td>Meters</td>
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</tbody>
</table>

When totals or summations are required the English values are summed separately from the Metric values. Differences may occur when comparing the English total with metric totals due to rounding.

All metric values are converted from English values. When using the conversion factors in this manual, all resultants will be rounded except when the value is a weight limitation. For minimum or maximum weight limitations the resultant metric values will be rounded up or truncated, whichever is more conservative.
2.0 AIRPLANE DESCRIPTION

2.1 GENERAL CHARACTERISTICS

Maximum Design Taxi Weight (MTW). Maximum weight for ground maneuver as limited by aircraft strength and airworthiness requirements. (It includes weight of taxi and run-up fuel.)

Maximum Design Takeoff Weight (MTOW). Maximum weight for takeoff as limited by aircraft strength and airworthiness requirements. (This is the maximum weight at start of the takeoff run.)

Maximum Design Landing Weight (MLW). Maximum weight for landing as limited by aircraft strength and airworthiness requirements.

Maximum Design Zero Fuel Weight (MZFW). Maximum weight allowed before usable fuel and other specified usable agents must be loaded in defined sections of the aircraft as limited by strength and airworthiness requirements.

Operating Empty Weight (OEW). Weight of structure, powerplant, furnishing systems, unusable fuel and other unusable propulsion agents, and other items of equipment that are considered an integral part of a particular airplane configuration. Also included are certain standard items, personnel, equipment, and supplies necessary for full operations, excluding usable fuel and payload.

Maximum Structural Payload. Maximum design zero fuel weight minus operating empty weight.

Maximum Seating Capacity. The maximum number of passengers specifically certificated or anticipated for certification.

Maximum Cargo Volume. The maximum space available for cargo.

Usable Fuel. Fuel available for aircraft propulsion.
2.1.1 General Characteristics: Model 787-8

<table>
<thead>
<tr>
<th>CHARACTERISTICS</th>
<th>UNITS</th>
<th>MODEL 787-8 *[1]</th>
</tr>
</thead>
<tbody>
<tr>
<td>MAX DESIGN TAXI WEIGHT</td>
<td>POUNDS</td>
<td>503,500</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>228,383</td>
</tr>
<tr>
<td>MAX DESIGN TAKEOFF WEIGHT</td>
<td>POUNDS</td>
<td>502,500</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>227,930</td>
</tr>
<tr>
<td>MAX DESIGN LANDING WEIGHT</td>
<td>POUNDS</td>
<td>380,000</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>172,365</td>
</tr>
<tr>
<td>MAX DESIGN ZERO FUEL WEIGHT</td>
<td>POUNDS</td>
<td>355,000</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>161,025</td>
</tr>
<tr>
<td>SEATING CAPACITY</td>
<td>ONE CLASS</td>
<td>359 ALL-ECONOMY SEATS; FAA EXIT LIMIT = 381 SEATS</td>
</tr>
<tr>
<td></td>
<td>MIXED CLASS</td>
<td>242 DUAL-CLASS; 24 BUSINESS CLASS, 218 ECONOMY CLASS (SEE SEC 2.4)</td>
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<tr>
<td>MAX CARGO - LOWER DECK *[2]</td>
<td>CUBIC FEET</td>
<td>4,826</td>
</tr>
<tr>
<td></td>
<td>CUBIC METERS</td>
<td>136.7</td>
</tr>
<tr>
<td>USABLE FUEL *[3]</td>
<td>U.S. GALLONS</td>
<td>33,340</td>
</tr>
<tr>
<td></td>
<td>LITERS</td>
<td>126,206</td>
</tr>
<tr>
<td></td>
<td>POUNDS</td>
<td>223,378</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>101,343</td>
</tr>
</tbody>
</table>

NOTES:

*[1] GENERAL CHARACTERISTICS ARE THE SAME FOR BOTH GENERAL ELECTRIC AND ROLLS ROYCE ENGINES.

*[2] 16 LD-3 CONTAINERS IN FWD COMPARTMENT AT 158 CU FT (4.5 CU M) EACH; 12 LD-3 CONTAINERS IN AFT COMPARTMENT; 402 CU FT (11.4 CU M) IN BULK CARGO COMPARTMENT. SEE SEC 2.6 FOR OTHER LOADING COMBINATIONS.

*[3] FUEL DENSITY = 6.7 LBS/US GAL
### 2.1.2 General Characteristics: Model 787-9

<table>
<thead>
<tr>
<th>CHARACTERISTICS</th>
<th>UNITS</th>
<th>MODEL 787-9 *[1]</th>
</tr>
</thead>
<tbody>
<tr>
<td>MAX DESIGN TAXI WEIGHT</td>
<td>POUNDS</td>
<td>563,000</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>255,372</td>
</tr>
<tr>
<td>MAX DESIGN TAKEOFF WEIGHT</td>
<td>POUNDS</td>
<td>561,500</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>254,692</td>
</tr>
<tr>
<td>MAX DESIGN LANDING WEIGHT</td>
<td>POUNDS</td>
<td>425,000</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>192,776</td>
</tr>
<tr>
<td>MAX DESIGN ZERO FUEL WEIGHT</td>
<td>POUNDS</td>
<td>400,000</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>181,436</td>
</tr>
<tr>
<td>SEATING CAPACITY</td>
<td>ONE CLASS</td>
<td>406 ALL-ECONOMY SEATS; FAA EXIT LIMIT = 420 SEATS</td>
</tr>
<tr>
<td></td>
<td>MIXED CLASS</td>
<td>290 DUAL-CLASS; 28 BUSINESS CLASS, 262 ECONOMY CLASS (SEE SEC 2.4)</td>
</tr>
<tr>
<td>MAX CARGO - LOWER DECK *[2]</td>
<td>CUBIC FEET</td>
<td>6,090</td>
</tr>
<tr>
<td></td>
<td>CUBIC METERS</td>
<td>172.4</td>
</tr>
<tr>
<td>USABLE FUEL *[3]</td>
<td>U.S. GALLONS</td>
<td>33,399</td>
</tr>
<tr>
<td></td>
<td>LITERS</td>
<td>126,429</td>
</tr>
<tr>
<td></td>
<td>POUNDS</td>
<td>223,773</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>101,522</td>
</tr>
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**NOTES:**

*[1] GENERAL CHARACTERISTICS ARE THE SAME FOR BOTH GENERAL ELECTRIC AND ROLLS ROYCE ENGINES.

*[2] 20 LD-3 CONTAINERS IN FWD COMPARTMENT AT 158 CU FT (4.5 CU M) EACH; 16 LD-3 CONTAINERS IN AFT COMPARTMENT; 402 CU FT (11.4 CU M) IN BULK CARGO COMPARTMENT. SEE SEC 2.6 FOR OTHER LOADING COMBINATIONS.

*[3] FUEL DENSITY = 6.7 LBS/US GAL
### 2.1.3 General Characteristics: Model 787-10

<table>
<thead>
<tr>
<th>CHARACTERISTICS</th>
<th>UNITS</th>
<th>MODEL 787-10 *[1]</th>
</tr>
</thead>
<tbody>
<tr>
<td>MAX DESIGN TAXI WEIGHT</td>
<td>POUNDS</td>
<td>561,500</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>254,692</td>
</tr>
<tr>
<td>MAX DESIGN TAKEOFF WEIGHT</td>
<td>POUNDS</td>
<td>560,000</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>254,011</td>
</tr>
<tr>
<td>MAX DESIGN LANDING WEIGHT</td>
<td>POUNDS</td>
<td>445,000</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>201,848</td>
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<td>MAX DESIGN ZERO FUEL WEIGHT</td>
<td>POUNDS</td>
<td>425,000</td>
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<tr>
<td></td>
<td>KILOGRAMS</td>
<td>192,776</td>
</tr>
<tr>
<td>SEATING CAPACITY</td>
<td>ONE CLASS</td>
<td>440 all-economy seats; FAA exit limit = 440 seats</td>
</tr>
<tr>
<td></td>
<td>MIXED CLASS</td>
<td>330 dual-class; 32 business class, 298 economy class (see sec 2.4)</td>
</tr>
<tr>
<td>MAX CARGO - LOWER DECK *[2]</td>
<td>CUBIC FEET</td>
<td>6,722</td>
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<tr>
<td></td>
<td>CUBIC METERS</td>
<td>190.3</td>
</tr>
<tr>
<td>USABLE FUEL *[3]</td>
<td>U.S. GALLONS</td>
<td>33,399</td>
</tr>
<tr>
<td></td>
<td>LITERS</td>
<td>126,429</td>
</tr>
<tr>
<td></td>
<td>POUNDS</td>
<td>223,773</td>
</tr>
<tr>
<td></td>
<td>KILOGRAMS</td>
<td>101,522</td>
</tr>
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</table>

**NOTES:**

*[1] GENERAL CHARACTERISTICS ARE THE SAME FOR BOTH GENERAL ELECTRIC AND ROLLS ROYCE ENGINES.

*[2] 22 LD-3 containers in fwd compartment at 158 cu ft (4.5 cu m) each; 18 LD-3 containers in aft compartment; 402 cu ft (11.4 cu m) in bulk cargo compartment. See sec 2.6 for other loading combinations.

*[3] FUEL DENSITY = 6.7 lbs/us gal
2.2 GENERAL DIMENSIONS

2.2.1 General Dimensions: Model 787-8

![Diagram of Model 787-8 dimensions with measurements and scales.]
2.2.2 General Dimensions: Model 787-9

- 133 FT 6 IN (40.69 M)
- 142 FT 3 IN (43.36 M)
- 206 FT 1 IN (62.81 M)
- 68 FT 3 IN (20.80 M)
- 32 FT 6 IN (9.91 M)
- 18 FT 11 IN (5.77 M)
- 19 FT 6 IN (5.94 M)
- 55 FT 10 IN (17.02 M)
- 17 FT 9 IN (5.41 M)
- 84 FT 9 IN (25.83 M)
- 203 FT 5 IN (62.00 M)
- 197 FT 3 IN (60.12 M)
- 65 FT 0 IN (19.81 M)
- 32 FT 2 IN (9.80 M)

**SCALE**

FEET METERS

0 2 4 6 8 10 12

0 10 20 30 40
2.2.3 General Dimensions: Model 787-10

- 143 ft 6 in (43.35 m)
- 152 ft 3 in (46.31 m)
- 224 ft 1 in (68.30 m)
- 78 ft 3 in (23.85 m)
- 31 ft 10 in (9.71 m)
- 18 ft 11 in (5.77 m)
- 19 ft 6 in (5.94 m)
- 55 ft 10 in (17.02 m)
- 94 ft 9 in (28.88 m)
- 17 ft 9 in (5.41 m)
- 221 ft 5 in (67.48 m)
- 197 ft 3 in (60.12 m)
- 65 ft 0 in (19.81 m)
- 32 ft 2 in (9.80 m)
2.3 GROUND CLEARANCES

2.3.1 Ground Clearances: Model 787-8

<table>
<thead>
<tr>
<th>Dimension</th>
<th>MINIMUM</th>
<th>MAXIMUM</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>FT - IN</td>
<td>M</td>
</tr>
<tr>
<td>A</td>
<td>25 – 2</td>
<td>7.67</td>
</tr>
<tr>
<td>B</td>
<td>13 – 11</td>
<td>4.24</td>
</tr>
<tr>
<td>C</td>
<td>7 – 9</td>
<td>2.36</td>
</tr>
<tr>
<td>D</td>
<td>5 – 6</td>
<td>1.68</td>
</tr>
<tr>
<td>E</td>
<td>14 – 5</td>
<td>4.39</td>
</tr>
<tr>
<td>F (GE ENGINES)</td>
<td>2 – 5</td>
<td>0.74</td>
</tr>
<tr>
<td>F (RR ENGINES)</td>
<td>2 – 4</td>
<td>0.71</td>
</tr>
<tr>
<td>G</td>
<td>15 – 1</td>
<td>4.60</td>
</tr>
<tr>
<td>H</td>
<td>8 – 9</td>
<td>2.67</td>
</tr>
<tr>
<td>J</td>
<td>23 – 10</td>
<td>7.26</td>
</tr>
<tr>
<td>K</td>
<td>8 – 11</td>
<td>2.72</td>
</tr>
<tr>
<td>L</td>
<td>15 – 3</td>
<td>4.65</td>
</tr>
<tr>
<td>M</td>
<td>22 – 3</td>
<td>6.78</td>
</tr>
<tr>
<td>N</td>
<td>54 – 5</td>
<td>16.59</td>
</tr>
</tbody>
</table>

NOTES:
1. VERTICAL CLEARANCES SHOWN OCCUR DURING MAXIMUM VARIATIONS OF AIRPLANE ATTITUDE. COMBINATIONS OF AIRPLANE LOADING AND UNLOADING ACTIVITIES THAT PRODUCE THE GREATEST POSSIBLE VARIATION IN ATTITUDE WERE USED TO ESTABLISH THE VARIATIONS SHOWN.
2. DURING ROUTINE SERVICING, THE AIRPLANE REMAINS RELATIVELY STABLE, PITCH AND ELEVATION CHANGES OCCURRING SLOWLY.

* NOMINAL DIMENSIONS ROUNDED TO NEAREST INCH AND NEAREST CENTIMETER.
## 2.3.2 Ground Clearances: Model 787-9

![Diagram of airplane with clearances marked]

<table>
<thead>
<tr>
<th>Dimension</th>
<th>MINIMUM</th>
<th>MAXIMUM</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>FT - IN</td>
<td>M</td>
</tr>
<tr>
<td>A</td>
<td>24 – 4</td>
<td>7.42</td>
</tr>
<tr>
<td>B</td>
<td>13 – 11</td>
<td>4.24</td>
</tr>
<tr>
<td>C</td>
<td>7 – 8</td>
<td>2.34</td>
</tr>
<tr>
<td>D</td>
<td>5 – 9</td>
<td>1.75</td>
</tr>
<tr>
<td>E</td>
<td>14 – 6</td>
<td>4.42</td>
</tr>
<tr>
<td>F (GE ENGINES)</td>
<td>2 – 3</td>
<td>0.69</td>
</tr>
<tr>
<td>F (RR ENGINES)</td>
<td>2 – 0</td>
<td>0.61</td>
</tr>
<tr>
<td>G</td>
<td>15 – 2</td>
<td>4.62</td>
</tr>
<tr>
<td>H</td>
<td>8 – 10</td>
<td>2.69</td>
</tr>
<tr>
<td>J</td>
<td>24 – 0</td>
<td>7.32</td>
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<tr>
<td>K</td>
<td>9 – 0</td>
<td>2.74</td>
</tr>
<tr>
<td>L</td>
<td>15 – 5</td>
<td>4.70</td>
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<tr>
<td>M</td>
<td>22 – 7</td>
<td>6.88</td>
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<td>N</td>
<td>55 – 2</td>
<td>16.81</td>
</tr>
</tbody>
</table>

**NOTES:**

1. VERTICAL CLEARANCES SHOWN OCCUR DURING MAXIMUM VARIATIONS OF AIRPLANE ATTITUDE. COMBINATIONS OF AIRPLANE LOADING AND UNLOADING ACTIVITIES THAT PRODUCE THE GREATEST POSSIBLE VARIATION IN ATTITUDE WERE USED TO ESTABLISH THE VARIATIONS SHOWN.

2. DURING ROUTINE SERVICING, THE AIRPLANE REMAINS RELATIVELY STABLE, PITCH AND ELEVATION CHANGES OCCURRING SLOWLY.

* NOMINAL DIMENSIONS ROUNDED TO NEAREST INCH AND NEAREST CENTIMETER.
### 2.3.3 Ground Clearances: Model 787-10

<table>
<thead>
<tr>
<th>Dimension</th>
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<th>MAXIMUM</th>
</tr>
</thead>
<tbody>
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<td>FT - IN</td>
<td>M</td>
</tr>
<tr>
<td>A</td>
<td>26-0</td>
<td>7.92</td>
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<tr>
<td>B</td>
<td>14-0</td>
<td>4.27</td>
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<tr>
<td>C</td>
<td>7-9</td>
<td>2.36</td>
</tr>
<tr>
<td>D</td>
<td>6-6</td>
<td>1.98</td>
</tr>
<tr>
<td>E</td>
<td>14-8</td>
<td>4.47</td>
</tr>
<tr>
<td>F (GE ENGINES)</td>
<td>2-5</td>
<td>0.74</td>
</tr>
<tr>
<td>F (RR ENGINES)</td>
<td>2-5</td>
<td>0.74</td>
</tr>
<tr>
<td>G</td>
<td>15-4</td>
<td>4.67</td>
</tr>
<tr>
<td>H</td>
<td>9-2</td>
<td>2.79</td>
</tr>
<tr>
<td>J</td>
<td>25-4</td>
<td>7.72</td>
</tr>
<tr>
<td>K</td>
<td>9-6</td>
<td>2.90</td>
</tr>
<tr>
<td>L</td>
<td>15-10</td>
<td>4.83</td>
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<tr>
<td>M</td>
<td>23-3</td>
<td>7.09</td>
</tr>
<tr>
<td>N</td>
<td>55-5</td>
<td>16.89</td>
</tr>
</tbody>
</table>

**NOTES:**

1. VERTICAL CLEARANCES SHOWN OCCUR DURING MAXIMUM VARIATIONS OF AIRPLANE ATTITUDE. COMBINATIONS OF AIRPLANE LOADING AND UNLOADING ACTIVITIES THAT PRODUCE THE GREATEST POSSIBLE VARIATION IN ATTITUDE WERE USED TO ESTABLISH THE VARIATIONS SHOWN.

2. DURING ROUTINE SERVICING, THE AIRPLANE REMAINS RELATIVELY STABLE, PITCH AND ELEVATION CHANGES OCCURRING SLOWLY.

* NOMINAL DIMENSIONS ROUNDED TO NEAREST INCH AND NEAREST CENTIMETER.
2.4 INTERIOR ARRANGEMENTS

2.4.1 Interior Arrangements - Typical: Model 787-8, 787-9, 787-10
2.5 CABIN CROSS SECTIONS

2.5.1 Cabin Cross-Sections: Model 787-8, 787-9, 787-10

FIRST CLASS/BUSINESS CLASS SEATS

9-ABREAST ECONOMY SEATS
2.6 LOWER CARGO COMPARTMENTS

2.6.1 Lower Cargo Compartments: Model 787-8, Containers and Bulk Cargo
2.6.2 Lower Cargo Compartments: Model 787-9, Containers and Bulk Cargo
2.6.3 Lower Cargo Compartments: Model 787-10, Containers and Bulk Cargo
2.7 DOOR CLEARANCES

2.7.1 Door Locations: Model 787-8, 787-9, 787-10, Passenger and Cargo Doors

<table>
<thead>
<tr>
<th>Door Name</th>
<th>Door Location</th>
<th>787-8 FT-IN/M</th>
<th>787-9 FT-IN/M</th>
<th>787-10 FT-IN/M</th>
</tr>
</thead>
<tbody>
<tr>
<td>1 MAIN ENTRY/SERVICE DOOR NO 1 (2)</td>
<td>LEFT AND RIGHT</td>
<td>20-8 / 6.30</td>
<td>20-8 / 6.30</td>
<td>20-8 / 6.30</td>
</tr>
<tr>
<td>2 MAIN ENTRY/SERVICE DOOR NO 2 (2)</td>
<td>LEFT AND RIGHT</td>
<td>50-3 / 15.32</td>
<td>60-3 / 18.36</td>
<td>70-3 / 21.41</td>
</tr>
<tr>
<td>3 MAIN ENTRY/SERVICE DOOR NO 3 (2)</td>
<td>LEFT AND RIGHT</td>
<td>106-3 / 32.39</td>
<td>116-3 / 35.43</td>
<td>126-3 / 38.48</td>
</tr>
<tr>
<td>4 MAIN ENTRY/SERVICE DOOR NO 4 (2)</td>
<td>LEFT AND RIGHT</td>
<td>142-11 / 43.56</td>
<td>162-11 / 49.66</td>
<td>180-11 / 55.14</td>
</tr>
<tr>
<td>5 FORWARD CARGO DOOR</td>
<td>RIGHT</td>
<td>36-1 / 11.00</td>
<td>36-1 / 11.00</td>
<td>36-1 / 11.00</td>
</tr>
<tr>
<td>6 AFT CARGO DOOR</td>
<td>RIGHT</td>
<td>122-1 / 37.21</td>
<td>142-1 / 43.31</td>
<td>152-1 / 46.36</td>
</tr>
<tr>
<td>7 BULK CARGO DOOR</td>
<td>LEFT</td>
<td>136-8 / 41.66</td>
<td>156-8 / 47.75</td>
<td>174-8 / 53.24</td>
</tr>
</tbody>
</table>

NOTES:
* [1] ENTRY DOORS LEFTSIDE, SERVICE DOORS RIGHTSIDE
* [2] SEE SECTION 2.3 FOR DOOR SILL HEIGHTS
2.7.2 Door Clearances: Model 787-8, 787-9, 787-10, Main Deck Entry and Service Doors
2.7.3 Door Clearances: Model 787-8, 787-9, 787-10, Lower Deck Cargo Door (Forward & Aft)
2.7.4 Door Clearances: Model 787-8, 787-9, 787-10, Bulk Cargo Door
3.0 AIRPLANE PERFORMANCE

3.1 GENERAL INFORMATION

The graphs in Section 3.2 provide information on payload-range capability of the 787 airplane. To use these graphs, if the trip range and zero fuel weight (OEW + payload) are known, the approximate takeoff weight can be found, limited by maximum zero fuel weight, maximum design takeoff weight, or fuel capacity.

The graphs in Section 3.3 provide information on FAA/EASA takeoff runway length requirements with typical engines at different pressure altitudes. Maximum takeoff weights shown on the graphs are the heaviest for the particular airplane models with the corresponding engines. Standard day temperatures for pressure altitudes shown on the FAA/EASA takeoff graphs are given below:

<table>
<thead>
<tr>
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<th>STANDARD DAY TEMP</th>
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<td>14,000</td>
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The graphs in Section 3.4 provide information on landing runway length requirements for different airplane weights and airport altitudes. The maximum landing weights shown are the heaviest for the particular airplane model.
3.2 PAYLOAD/RANGE FOR LONG RANGE CRUISE

3.2.1 Payload/Range for Long-Range Cruise: Model 787-8 (Typical Engines)
3.2.2 Payload/Range for Long-Range Cruise: Model 787-9 (Typical Engines)
3.2.3 Payload/Range for Long-Range Cruise: Model 787-10
(Typical Engines)
3.3 FAA/EASA TAKEOFF RUNWAY LENGTH REQUIREMENTS

3.3.1 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-8 (Typical Engines)
3.3.2 FAA/EASA Take Off Runway Length Requirements - Standard Day
+27°F (STD +15°C), Dry Runway: Model 787-8 (Typical Engines)
3.3.3 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-8 (Typical Engines)
3.3.4 FAA/EASA Takeoff Runway Length Requirements - Standard Day
+ 61°F (STD + 34°C, Dry Runway): Model 787-8 (Typical Engines)
3.3.5 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-8 (Hi-Thrust Engines)
3.3.6 FAA/EASA Takeoff Runway Length Requirements - Standard Day
+ 27°F (STD + 15°C), Dry Runway: Model 787-8 (Hi-Thrust Engines)
3.3.8 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 61°F (STD + 34°C), Dry Runway: Model 787-8 (Hi-Thrust Engines)
3.3.9 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-9 (Typical Engines)
3.3.11 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-9 (Typical Engines)
3.3.12 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 61°F (STD + 34°C), Dry Runway: Model 787-9 (Typical Engines)
3.3.13 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-9, (Hi-Thrust Engines)
3.3.14 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 27°F (STD + 15°C), Dry Runway: Model 787-9, (Hi-Thrust Engines)
3.3.15 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-9, (Hi-Thrust Engines)
3.3.16 FAA/EASA Takeoff Runway Length Requirements - Standard Day +61°F (STD +34°C), Dry Runway; Model 787-9 (Hi-Thrust Engines)
3.3.17 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-10 (Typical Engines)
3.3.18 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 27°F (STD + 15°C), Dry Runway: Model 787-10 (Typical Engines)
3.3.19 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 45°F (STD + 25°C), Dry Runway: Model 787-10 (Typical Engines)
3.3.20 FAA/EASA Takeoff Runway Length Requirements - Standard Day
+ 61°F (STD + 34°C), Dry Runway: Model 787-10 (Typical Engines)
3.3.21 FAA/EASA Takeoff Runway Length Requirements - Standard Day, Dry Runway: Model 787-10 (Hi-Thrust Engines)
3.3.22 FAA/EASA Takeoff Runway Length Requirements - Standard Day + 27°F (STD + 15°C), Dry Runway: Model 787-10 (Hi-Thrust Engines)
3.3.23 FAA/EASA Takeoff Runway Length Requirements - Standard Day
+ 45°F (STD + 25°C), Dry Runway: Model 787-10 (Hi-Thrust Engines)
3.3.24 FAA/EASA Takeoff Runway Length Requirements - Standard Day
+ 61°F (STD + 34°C), Dry Runway: Model 787-10 (Hi-Thrust Engines)
3.4 FAA/EASA LANDING RUNWAY LENGTH REQUIREMENTS

3.4.1 FAA/EASA Landing Runway Length Requirements – Flaps 30: Model 787-8 (All Engines)
3.4.2 FAA/EASA Landing Runway Length Requirements – Flaps 25: Model 787-8 (All Engines)
3.4.3 FAA/EASA Landing Runway Length Requirements – Flaps 30: Model 787-9 (All Engines)
3.4.4 FAA/EASA Landing Runway Length Requirements – Flaps 25: Model 787-9 (All Engines)
3.4.5 FAA/EASA Landing Runway Length Requirements – Flaps 30: Model 787-10 (All Engines)
3.4.6  FAA/EASA Landing Runway Length Requirements – Flaps 25: Model 787-10 (All Engines)
4.0 GROUND MANEUVERING

4.1 GENERAL INFORMATION

This section provides airplane turning capability and maneuvering characteristics.

For ease of presentation, these data have been determined from the theoretical limits imposed by the geometry of the aircraft, and where noted, provide for a normal allowance for tire slippage. As such, they reflect the turning capability of the aircraft in favorable operating circumstances. These data should be used only as guidelines for the method of determination of such parameters and for the maneuvering characteristics of this aircraft.

In the ground operating mode, varying airline practices may demand that more conservative turning procedures be adopted to avoid excessive tire wear and reduce possible maintenance problems. Airline operating procedures will vary in the level of performance over a wide range of operating circumstances throughout the world. Variations from standard aircraft operating patterns may be necessary to satisfy physical constraints within the maneuvering area, such as adverse grades, limited area, or high risk of jet blast damage. For these reasons, ground maneuvering requirements should be coordinated with the using airlines prior to layout planning.

Section 4.2 presents turning radii for various nose gear steering angles. Radii for the main and nose gears are measured from the turn center to the outside of the tire.

Section 4.3 shows data on minimum width of pavement required for 180° turn.

Section 4.4 provides pilot visibility data from the cockpit and the limits of ambinocular vision through the windows. Ambinocular vision is defined as the total field of vision seen simultaneously by both eyes.

Section 4.5 shows approximate wheel paths for various runway and taxiway turn scenarios. The pavement fillet geometries are based on the FAA’s Advisory Circular (AC) 150/5300-13 (thru change 16). They represent typical fillet geometries built at many airports worldwide. ICAO and other civil aviation authorities publish many different fillet design methods. Prior to determining the size of fillets, airports are advised to check with the airlines regarding the operating procedures and aircraft types they expect to use at the airport. Further, given the cost of modifying fillets and the operational impact to ground movement and air traffic during construction, airports may want to design critical fillets for larger aircraft types to minimize future operational impacts.

Section 4.6 illustrates a typical runway holding bay configuration.
4.2 TURNING RADII

4.2.1 Turning Radii – No Slip Angle: Model 787-8

NOTES: ACTUAL OPERATING TURNING RADII MAY BE GREATER THAN SHOWN. CONSULT WITH AIRLINE FOR SPECIFIC OPERATING PROCEDURE.

<table>
<thead>
<tr>
<th>STEERING ANGLE</th>
<th>R1 INNER GEAR</th>
<th>R2 OUTER GEAR</th>
<th>R3 NOSE GEAR</th>
<th>R4 WINGTIP</th>
<th>R5 NOSE</th>
<th>R6 TAIL</th>
</tr>
</thead>
<tbody>
<tr>
<td>(DEG)</td>
<td>FT  M</td>
<td>FT  M</td>
<td>FT  M</td>
<td>FT  M</td>
<td>FT  M</td>
<td>FT  M</td>
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<tr>
<td>30</td>
<td>111 33.8</td>
<td>149 45.4</td>
<td>152 46.3</td>
<td>232 70.7</td>
<td>160 48.8</td>
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<tr>
<td>35</td>
<td>88 26.8</td>
<td>126 38.4</td>
<td>133 40.5</td>
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<tr>
<td>40</td>
<td>71 21.6</td>
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<td>129 39.3</td>
<td>154 46.9</td>
</tr>
<tr>
<td>45</td>
<td>56 17.1</td>
<td>94 28.7</td>
<td>108 32.9</td>
<td>178 54.3</td>
<td>119 36.3</td>
<td>143 43.6</td>
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<tr>
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<td>44 13.4</td>
<td>82 25.0</td>
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<tr>
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<td>34 10.4</td>
<td>72 21.9</td>
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<td>107 32.6</td>
<td>127 38.7</td>
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<td>82 25.0</td>
<td>133 40.5</td>
<td>97 29.6</td>
<td>112 34.1</td>
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### 4.2.2 Turning Radii – No Slip Angle: Model 787-9

**NOTES:** ACTUAL OPERATING TURNING RADII MAY BE GREATER THAN SHOWN. CONSULT WITH AIRLINE FOR SPECIFIC OPERATING PROCEDURE.

<table>
<thead>
<tr>
<th>STEERING ANGLE</th>
<th>R1 INNER GEAR</th>
<th>R2 OUTER GEAR</th>
<th>R3 NOSE GEAR</th>
<th>R4 WINGTIP</th>
<th>R5 NOSE</th>
<th>R6 TAIL</th>
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<tbody>
<tr>
<td>(DEG)</td>
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<td>FT M</td>
<td>FT M</td>
<td>FT M</td>
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<td>169 51.5</td>
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<tr>
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<td>105 32.0</td>
<td>122 37.2</td>
<td>188 57.3</td>
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<td>147 44.8</td>
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<td>79 24.1</td>
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<td>108 32.9</td>
<td>122 37.2</td>
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### 4.2.3 Turning Radii – No Slip Angle: Model 787-10

**NOTES:** ACTUAL OPERATING TURNING RADII MAY BE GREATER THAN SHOWN. CONSULT WITH AIRLINE FOR SPECIFIC OPERATING PROCEDURE.

<table>
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<th>STEERING ANGLE (DEG)</th>
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<th>R2 OUTER GEAR</th>
<th>R3 NOSE GEAR</th>
<th>R4 WINGTIP</th>
<th>R5 NOSE</th>
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REV O February 2023 4-4

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4.3 CLEARANCE RADII: MODEL 787-8, 787-9, 787-10

**Table:**

<table>
<thead>
<tr>
<th>MODEL</th>
<th>EFFECTIVE TURNING ANGLE (DEG)</th>
<th>X</th>
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<td>155</td>
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<td>47</td>
<td>14.3</td>
<td>173</td>
<td>52.7</td>
<td>108</td>
</tr>
</tbody>
</table>

THEORETICAL CENTER OF TURN FOR MINIMUM TURNING RADIUS SLOW CONTINUOUS TURNING AT MINIMUM THRUST ON ALL ENGINES. NO DIFFERENTIAL BRAKING. CONSULT USING AIRLINE FOR SPECIFIC OPERATING PROCEDURES.
4.4 VISIBILITY FROM COCKPIT IN STATIC POSITION

4.4.1 Visibility from Cockpit in Static Position: Model 787-8, 787-9, 787-10

NOT TO SCALE

PILOT EYE POSITION

NOT TO BE USED FOR LANDING APPROACH VISIBILITY

ABOVE GROUND

[787-8] 17 FT 9 IN (5.41 m)
[787-9] 18 FT 0 IN (5.48 m)
[787-10] 19 FT 2 IN (5.54 m)

TAXI ATTITUDE

[787-8] +1.0°
[787-9] -0.52°
[787-10] -0.52°

[ALL MODELS] 25.4°(1)

[ALL MODELS] 21.0°(2)

[ALL MODELS] 8 FT 8 IN (2.64 m)

[787-8] 43 FT 11 IN (13.38 m)
[787-9] 45 FT 7 IN (13.89 m)
[787-10] 46 FT 1 IN (14.09 m)

VISUAL ANGLES IN VERTICAL PLANE THROUGH PILOT'S EYE POSITION

[ALL MODELS] 21 IN (0.53 m)

[ALL MODELS] 121.5°(3)(4)

VISUAL ANGLES IN HORIZONTAL PLANE

NOTES:

(1) UPWARD THROUGH #1 WINDOW
(2) DOWNWARD THROUGH #1 WINDOW
(3) VISION THROUGH #2 WINDOW
(4) HEAD ROTATED ABOUT POINT 3.3 IN (0.08 m) AFT OF PILOT'S REFERENCE EYE POSITION.
4.5 RUNWAY AND TAXIWAY TURNPATHS

4.5.1 Runway and Taxiway Turnpaths - Runway-to-Taxiway, More Than 90 Degree Turn: Model 787-8

NOTE
BEFORE DETERMINING THE SIZE OF THE INTERSECTION FILLET
CHECK WITH THE AIRLINES REGARDING THE OPERATING PROCEDURES
IN USE AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT
4.5.2 Runway and Taxiway Turnpaths - Runway-to-Taxiway, 90 Degree Turn: Model 787-8

NOTE

BEFORE DETERMINING THE SIZE OF THE INTERSECTION FILLET
CHECK WITH THE AIRLINES REGARDING THE OPERATING PROCEDURES
IN USE AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT

---

Cockpit tracks
Centerline of turns
4.5.3 Runway and Taxiway Turnpaths - Taxiway-to-Taxiway, 90 Degree Turn: Model 787-8

NOTE

BEFORE DETERMINING THE SIZE OF THE INTERSECTION FILLET
CHECK WITH THE AIRLINES REGARDING THE OPERATING PROCEDURES
IN USE AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT

[Diagram showing runway and taxiway turnpaths with labels such as APPROX NOSE GEAR PATH, APPROX 25 FT (8 M), R= 85 FT (26 M), FAA LEAD-IN FILLET, TRACK OF OUTSIDE EDGE OF OUTBOARD WHEEL, CENTERLINE OF TAXIWAY, and COCKPIT TRACKS CENTERLINE OF TURNS.]
4.5.4 Runway and Taxiway Turnpaths - Runway-to-Taxiway, More Than 90 Degree Turn: Model 787-9

NOTE
BEFORE DETERMINING THE SIZE OF THE INTERSECTION FILLET
CHECK WITH THE AIRLINES REGARDING THE OPERATING PROCEDURES
IN USE AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT.
4.5.5 Runway and Taxiway Turnpaths - Runway-to-Taxiway, 90 Degree Turn: Model 787-9

NOTE

BEFORE DETERMINING THE SIZE OF THE INTERSECTION FILLET
CHECK WITH THE AIRLINES REGARDING THE OPERATING PROCEDURES
IN USE AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT

R=150 FT (45 M)
APPROX NOSE GEAR PATH

R=85 FT (26 M)
FAA LEAD-IN FILLET

APPROX 22 FT (7 M)

TRACK OF OUTSIDE EDGE OF OUTBOARD WHEEL
CENTERLINE OF RUNWAY

150 FT (45 M)

COCKPIT TRACKS
CENTERLINE OF TURNS
4.5.6 Runway and Taxiway Turnpaths - Taxiway-to-Taxiway, 90 Degree Turn: Model 787-9

NOTE
BEFORE DETERMINING THE SIZE OF THE INTERSECTION FILLET
CHECK WITH THE AIRLINES REGARDING THE OPERATING PROCEDURES
IN USE AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT

![Diagram of taxiway turnpaths](image-url)
4.5.7 Runway and Taxiway Turnpaths - Runway-to-Taxiway, More Than 90 Degree Turn: Model 787-10

NOTE

BEFORE DETERMINING THE SIZE OF THE INTERSECTION FILLET
CHECK WITH THE AIRLINES REGARDING THE OPERATING PROCEDURES
IN USE AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT

APPROX NOSE GEAR PATH

R=150 FT (45 M)

APPROX 11 FT (3 M)

R=80 FT (24 M)

TRACK OF OUTSIDE EDGE OF OUTBOARD WHEEL

FAA LEAD-IN FILLET

75 FT (23 M)

CENTERLINE OF RUNWAY

150 FT (45 M)

COCKPIT TRACKS
CENTERLINE OF TURNS
4.5.8 Runway and Taxiway Turnpaths - Runway-to-Taxiway, 90 Degree Turn: Model 787-10

NOTE

BEFORE DETERMINING THE SIZE OF THE INTERSECTION FILLET
CHECK WITH THE AIRLINES REGARDING THE OPERATING PROCEDURES
IN USE AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT
4.5.9 Runway and Taxiway Turnpaths - Taxiway-to-Taxiway, 90 Degree Turn: Model 787-10

NOTE

BEFORE DETERMINING THE SIZE OF THE INTERSECTION FILLET
CHECK WITH THE AIRLINES REGARDING THE OPERATING PROCEDURES
IN USE AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT

---

Diagram showing:
- R=150 FT (45 M)
- APPROX 15 FT (5M)
- R=85 FT (26 M)
- APPROX NOSE GEAR PATH
- TRACK OF OUTSIDE EDGE OF OUTBOARD WHEEL
- CENTERLINE OF TAXIWAY
- FAA LEAD-IN FILLET

Text: COCKPIT TRACKS, CENTERLINE OF TURNS

---
4.6 RUNWAY HOLDING BAY: MODEL 787-8, 787-9, 787-10

SHOULDER

20 FT (6 M)

APPROX 309 FT (94 M)

TO RUNWAY

150 FT (45 M)

40 FT (12 M)

20 FT (6 M) CLEARANCE BETWEEN CENTERLINE OF GEAR AND PAVEMENT EDGE

75 FT (23 M)

NOTE
BEFORE DETERMINING THE SIZE OF THE CORNER FILLET, CHECK WITH THE AIRLINES REGARDING OPERATING PROCEDURES AND THE AIRCRAFT TYPES EXPECTED TO SERVE THE AIRPORT.
5.0 TERMINAL SERVICING

During turnaround at the terminal, certain services must be performed on the aircraft, usually within a given time, to meet flight schedules. This section shows service vehicle arrangements, schedules, locations of service points, and typical service requirements. The data presented in this section reflect ideal conditions for a single airplane. Service requirements may vary according to airplane condition and airline procedure.

Section 5.1 shows typical arrangements of ground support equipment during turnaround. As noted, if the auxiliary power unit (APU) is used, the electrical, air start, and air-conditioning service vehicles would not be required. Passenger loading bridges or portable passenger stairs could be used to load or unload passengers.

Sections 5.2 and 5.3 show typical service times at the terminal. These charts give typical schedules for performing service on the airplane within a given time. Service times may be rearranged to suit availability of personnel, airplane configuration, and degree of service required.

Section 5.4 shows the locations of ground service connections in graphic and in tabular forms. Typical capacities and service requirements are shown in the tables. Services with requirements that vary with conditions are described in subsequent sections.

Section 5.5 shows minimum electrical ground power requirements for engine start. The curves are based on 120-second and 180-second start times depending on the ground power unit.

Section 5.6 shows air conditioning requirements for heating and cooling (pull-down and pull-up) using ground conditioned air. The curves show airflow requirements to heat or cool the airplane within a given time at ambient conditions.

Section 5.7 shows air conditioning requirements for heating and cooling to maintain a constant cabin air temperature using low pressure conditioned air. This conditioned air is supplied through an 8-in (20.3 cm) ground air connection (GAC) directly to the passenger cabin, bypassing the air cycle machines.

Section 5.8 shows ground towing requirements for various ground surface conditions.
5.1 AIRPLANE SERVICING ARRANGEMENT - TYPICAL TURNAROUND

5.1.1 Airplane Servicing Arrangement - Typical Turnaround: Model 787-8
5.1.2 Airplane Servicing Arrangement - Typical Turnaround: Model 787-9
5.1.3 Airplane Servicing Arrangement - Typical Turnaround: Model 787-10
5.1.4 Airplane Servicing Arrangement - Typical En Route: Model 787-8
5.1.5 Airplane Servicing Arrangement - Typical En Route: Model 787-9
5.1.6  Airplane Servicing Arrangement - Typical En Route: Model 787-10
5.2 TERMINAL OPERATIONS - TURNAROUND STATION

5.2.1 Terminal Operations, Turntime Analysis - Turnaround Station: Model 787-8
5.2.2 Terminal Operations, Turntime Analysis - Turnaround Station: Model 787-9
### Terminal Operations, Turntime Analysis - Turnaround Station:

#### Model 787-10

<table>
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<tr>
<th>Activity</th>
<th>Time (Minutes)</th>
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<td>Service Potable Water</td>
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<td>Service Lavatories</td>
<td>10.0</td>
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<tr>
<td>Chocks Off/ Pushback</td>
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</table>

#### Parameters:
- 100% Passenger and Cargo Exchange
- 411 Passengers, 2 Classes, 1 Door
- Passenger deplane rate is 40 per minute
- Passenger boarding rate is 25 per minute
- Galley Service Trucks
- (1) Lavatory Service Truck
- (1) Potable Water Service Truck
- Cabin service is available time
- Unload and load bulk cargo is available time
- (18) Containers Aft
- (22) Containers Forward
- 29,854 gallons (112,253 liters) fuel loaded with 3,730 gallon (14,120 liters) reserve
- Nozzle hydrant fueling at 50 psig
5.3 TERMINAL OPERATIONS - EN ROUTE STATION

5.3.1 Terminal Operations, Turntime Analysis - En Route Station: Model 787-8
### 5.3.2 Terminal Operations, Turntime Analysis - En Route Station: Model 787-9

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<tr>
<td>Chocks Off / Pushback</td>
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</table>

**Parameters:**
- 80% passenger and cargo exchange
- 360 passengers, 1 doors, 2 classes
- No galley service
- No lavatory service
- No cabin service

- Unload and load bulk cargo is available time
- 12 containers aft
- 16 containers forward
- 60567 liters (16000 gallons) fuel loaded
- Nozzle hydrant fueling at 50 psig
- Bulk cargo in bulk compartment at 75% utilization & 8.5 lb/cu ft
5.4 GROUND SERVICE CONNECTIONS

5.4.1 Ground Service Connections: Model 787-8
### 5.4.2 Ground Service Connections and Capacities: Model 787-8

<table>
<thead>
<tr>
<th>SYSTEM</th>
<th>MODEL</th>
<th>DISTANCE AFT OF DISTANCE FROM AIRPLANE CENTERLINE</th>
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<td>M</td>
<td>FT</td>
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5.4.3 Ground Servicing Connections: Model 787-9
### 5.4.4 Ground Servicing Connections and Capacities: Model 787-9

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5.4.5 Ground Servicing Connections: Model 787-10
### 5.4.6 Ground Servicing Connections and Capacities: Model 787-10

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<tr>
<td>WITH TWO FUELING PORTS</td>
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<td>FUEL VENTS</td>
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<tr>
<td>TOTAL CAPACITY 33,380 US GAL (126,205 LITERS)</td>
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<td>23.7</td>
<td>78</td>
<td>23.7</td>
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</table>
Normal engine start for the 787 uses the APU to provide electrical power. If the APU is inoperative or unavailable, an engine start can be accomplished using a minimum of two 90 kVA external ground power units connected to the two forward external receptacles. Boeing recommends using three 90 kVA ground power sources to minimize the effect on cabin load shedding of ventilation, In Flight Entertainment, and cabin lighting.
Normal engine start for the 787 uses the APU to provide electrical power. If the APU is inoperative or unavailable, an engine start can be accomplished using a minimum of two 90 kVA external ground power units connected to the two forward external receptacles. Boeing recommends using three 90 kVA ground power sources to minimize the effect on cabin load shedding of ventilation, In Flight Entertainment, and cabin lighting.
5.5.3 Engine Power Requirements – Pneumatic: Model 787-8, 787-9, 787-10

The 787 aircraft is an electric aircraft and does not have a traditional pneumatic system onboard, thus there are no ground pneumatic connections.
5.6 CONDITIONED AIR REQUIREMENTS

5.6.1 Conditioned Air Flow Requirements – Cooling Time: Model 787-8

**CONDITIONS:**
- OUTSIDE AIR TEMPERATURE: 103°F (39.4°C)
- INITIAL CABIN TEMPERATURE: 115°F (46.1°C)
- ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
- "HEAT REDUCTION MODE" SELECTED VIA CABIN ATTENDANT PANEL TO DIM WINDOWS AND LIGHTING
- RECIRCULATION FANS SELECTED OFF
- ICS RECIRCULATION CHILLING OFF
- IFE OFF
- NO OCCUPANTS
- FULL SOLAR LOAD

**LEGEND:**
- PCA TEMPERATURE AT GROUND CONNECTION
  - 50°F (10°C)
  - 45°F (7.2°C)
  - 40°F (4.4°C)
  - 35°F (1.7°C)

**NOTE:**
This graph provides the predicted time required to cool the airplane's cabin to a bulk average of 75°F (24°C) as a function of airflow and temperature, at the ground air connection, when using a pre-conditioned air (PCA) source.
5.6.2 Conditioned Air Flow Requirements – Cooling – Steady State
(103°F Ambient Air): Model 787-8

**CONDITIONS:**
- OUTSIDE AIR TEMPERATURE: 103°F (39.4°C)
- ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
- "HEAT REDUCTION MODE" SELECTED VIA CABIN
- ATTENDANT PANEL TO DIM WINDOWS AND LIGHTING
- RECIRCULATION FANS SELECTED OFF
- ICS RECIRCULATION CHILLING OFF
- IFE OFF
- 100% OCCUPANT LOAD
- FULL SOLAR LOAD

**NOTE:**
THIS GRAPH PROVIDES THE FLOW RATE AND TEMPERATURE, AT THE AIRPLANE’S GROUND AIR CONNECTION, THAT IS REQUIRED TO MAINTAIN THE AIRPLANE’S CABIN AT A BULK AVERAGE OF 75°F (24°C) WHEN USING A PRE-CONDITIONED AIR (PCA) SOURCE.

**LEGEND:**
- 353 SEATS
- 284 SEATS
5.6.3 Conditioned Air Flow Requirements – Cooling – Steady State (80 F Ambient Air): Model 787-8

CONCLUSIONS:

- OUTSIDE AIR TEMPERATURE: 80°F (26.7°C)
- ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
- HEAT REDUCTION MODE SELECTED VIA CABIN ATTENDANT PANEL TO DIM WINDOWS AND LIGHTING
- RECIRCULATION FANS SELECTED OFF
- ICS RECIRCULATION CHILLING OFF
- IFE OFF
- 100% OCCUPANT LOAD
- FULL SOLAR LOAD

LEGEND:

- 353 SEATS
- 284 SEATS

NOTE:

THIS GRAPH PROVIDES THE FLOW RATE AND TEMPERATURE, AT THE AIRPLANE'S GROUND AIR CONNECTION, THAT IS REQUIRED TO MAINTAIN THE AIRPLANE'S CABIN AT A BULK AVERAGE OF 75°F (24°C) WHEN USING A PRE-CONDITIONED AIR (PCA) SOURCE.
5.6.4 Conditioned Air Flow Requirements – Heating Time: Model 787-8

![Graph showing conditioned air flow requirements for heating time.](image-url)

**CONDITIONS:**
- All exterior doors and windows are closed
- Outside air temperature is -40°F (-40°C)
- Initial cabin temperature is -25°F (-32°C)
- No solar load
- Recirculation fans selected off
- ICS recirculation chilling off
- No occupants
- LFE off
- No electrical loads

**LEGEND:**
- PCA temperatures at ground connection
  - 120°F (48.9°C)
  - 140°F (60.0°C)
  - 160°F (71.1°C)

**NOTE:**
This graph provides the predicted time required to heat the airplane’s cabin to a bulk average of 75°F (24°C) as a function of airflow and temperature, at the ground air connection, when using a pre-conditioned air (PCA) source.
5.6.5 Conditioned Air Flow Requirements – Heating – Steady State: Model 787-8

**CONDITIONS:**
- ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
- NO SOLAR HEAT LOAD
- RECIRCULATION FANS SELECTED OFF
- ICS RECIRCULATION CHILLING OFF
- NO OCCUPANTS
- NO ELECTRICAL HEAT LOADS

**NOTE:**
This graph provides the flow rate and temperature, at the ground air connection, that is required to heat the airplane's cabin to a bulk average of 75°F (24°C) when using a pre-conditioned air (PCA) source.
5.6.6 Conditioned Air Flow Requirements – Cooling Time: Model 787-9

CONDITIONS:
- OUTSIDE AIR TEMPERATURE: 103°F (39.4°C)
- INITIAL CABIN TEMPERATURE: 115°F (46.1°C)
- ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
- "HEAT REDUCTION MODE" SELECTED VIA CABIN ATTENDANT PANEL TO DIM WINDOWS AND LIGHTING
- RECIRCULATION FANS SELECTED OFF
- ICS RECIRCULATION CHILLING OFF
- IFE OFF
- NO OCCUPANTS
- FULL SOLAR LOAD

LEGEND:
PCA TEMPERATURE AT GROUND CONNECTION
- 50°F (10°C)
- 45°F (7.2°C)
- 40°F (4.4°C)
- 35°F (1.7°C)

NOTE:
THIS GRAPH PROVIDES THE PREDICTED TIME REQUIRED TO COOL THE AIRPLANE'S CABIN TO A BULK AVERAGE OF 75°F (24°C) AS A FUNCTION OF AIRFLOW AND TEMPERATURE, AT THE GROUND AIR CONNECTION, WHEN USING A PRE-CONDITIONED AIR (PCA) SOURCE.
5.6.7 Conditioned Air Flow Requirements – Cooling – Steady State
(103 F Ambient Air): Model 787-9

CONDITIONS:
• OUTSIDE AIR TEMPERATURE: 103°F (39.4°C)
• ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
• "HEAT REDUCTION MODE" SELECTED VIA CABIN ATTENDANT PANEL,
  DIMMED WINDOWS AND LIGHTING
• LOWER RECIRCULATION FANS SELECTED OFF (UPPER FAN ON)
• ICS RECIRCULATION CHILLING OFF
• IFE OFF
• 100% OCCUPANT LOAD FACTOR
• FULL SOLAR LOAD

NOTE:
THIS GRAPH PROVIDES THE FLOW RATE AND TEMPERATURE, AT THE AIRPLANE’S GROUND
AIR CONNECTION, THAT IS REQUIRED TO MAINTAIN THE AIRPLANE’S CABIN AT A BULK
AVERAGE OF 75°F (24°C) WHEN USING A PRE-CONDITIONED AIR (PCA) SOURCE.

LEGEND:
- 402 SEATS
- 330 SEATS
5.6.8 Conditioned Air Flow Requirements – Cooling – Steady State (80 F Ambient Air): Model 787-9

**CONDITIONS:**
- OUTSIDE AIR TEMPERATURE: 80°F (26.7°C)
- ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
- "HEAT REDUCTION MODE" SELECTED VIA CABIN ATTENDANT PANEL, DIMMED WINDOWS AND LIGHTING
- LOWER RECIRCULATION FANS SELECTED OFF (UPPER FAN ON)
- ICS RECIRCULATION CHILLING OFF
- IFE OFF
- 100% OCCUPANT LOAD FACTOR
- FULL SOLAR LOAD

**LEGEND:**
- 402 SEATS
- 330 SEATS

**NOTE:**
THIS GRAPH PROVIDES THE FLOW RATE AND TEMPERATURE, AT THE AIRPLANE'S GROUND AIR CONNECTION, THAT IS REQUIRED TO MAINTAIN THE AIRPLANE'S CABIN AT A BULK AVERAGE OF 75°F (24°C) WHEN USING A PRE-CONDITIONED AIR (PCA) SOURCE.
5.6.9 Conditioned Air Flow Requirements – Heating Time: Model 787-9

<table>
<thead>
<tr>
<th>TOTAL AIRFLOW AT GROUND AIR CONDITION -</th>
<th>TIME TO HEAT CABIN TO 75°F (24°C) - MINUTES</th>
</tr>
</thead>
<tbody>
<tr>
<td>650 (249.5)</td>
<td>120°F (48.9°C)</td>
</tr>
<tr>
<td>500 (229.5)</td>
<td>140°F (60°C)</td>
</tr>
<tr>
<td>450 (204.1)</td>
<td>160°F (71.1°C)</td>
</tr>
<tr>
<td>400 (181.4)</td>
<td></td>
</tr>
<tr>
<td>350 (158.6)</td>
<td></td>
</tr>
<tr>
<td>300 (136.1)</td>
<td></td>
</tr>
<tr>
<td>250 (113.4)</td>
<td></td>
</tr>
<tr>
<td>200 (90.7)</td>
<td></td>
</tr>
</tbody>
</table>

 CONDITIONS:
- ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
- OUTSIDE AIR TEMPERATURE IS -40°F (-40°C)
- INITIAL CABIN TEMPERATURE IS -25°F (-31.7°C)
- NO SOLAR HEAT LOAD
- LOWER RECIRCULATION FANS SELECTED OFF (UPPER FAN ON)
- ICS RECIRCULATION CHILLING OFF
- NO OCCUPANTS
- IFE OFF
- NO ELECTRICAL HEAT LOADS

 LEGEND:
- PCA TEMPERATURE AT GROUND CONNECTION
- 120°F (48.9°C)
- 140°F (60.0°C)
- 160°F (71.1°C)

 NOTE:
THIS GRAPH PROVIDES THE PREDICTED TIME REQUIRED TO HEAT THE AIRPLANE’S CABIN TO A BULK AVERAGE OF 75°F (24°C) AS A FUNCTION OF AIRFLOW AND TEMPERATURE, AT THE GROUND AIR CONNECTION, WHEN USING A PRE-CONDITIONED AIR (PCA) SOURCE.
5.6.10 Conditioned Air Flow Requirements – Heating – Steady State: Model 787-9

**CONDITIONS:**
- All exterior doors and windows are closed
- No solar heat load
- Lower recirculation fans selected off (Upper fan on)
- ICS recirculation chilling off
- No occupants
- IFE off
- No electrical heat loads

**LEGEND:**
- 40°F (−40°C) AMBIENT
- 0°F (−17.8°C) AMBIENT

**NOTE:**
This graph provides the flow rate and temperature, at the ground air connection, that is required to heat the airplane's cabin to a bulk average of 75°F (24°C) when using a pre-conditioned air (PCA) source.
5.6.11 Conditioned Air Flow Requirements – Cooling Time: Model 787-10

**CONDITIONS:**
- **OUTSIDE AIR TEMPERATURE:** 103° F (39.4° C)
- **INITIAL CABIN TEMPERATURE:** 115° F (46.1° C)
- **ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED**
- **'HEAT REDUCTION MODE' SELECTED VIA CABIN ATTENDANT PANEL (CAP) (FULL SOLAR LOAD WITH DIMMED WINDOWS AND LIGHTING)**
- **LOWER RECIRCULATION FANS SELECTED OFF (UPPER FAN ON)**
- **ICS RECIRCULATION CHILLING OFF**
- **IFE OFF**
- **NO OCCUPANTS**

**LEGEND:**
- **PCA TEMPERATURE AT GROUND CONNECTION**
  - 50° F (10° C)
  - 45° F (7.2° C)
  - 40° F (4.4° C)
  - 35° F (1.7° C)

**NOTE:**
This graph provides the predicted time required to cool the airplane’s cabin to 75° F (24° C) as a function of airflow and temperature, at the ground air connection, when using a pre-conditioned air (PCA) source.
5.6.12 Conditioned Air Flow Requirements – Cooling – Steady State
(103 °F Ambient Air): Model 787-10

CONDITIONS:
- ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
- 'HEAT REDUCTION MODE' SELECTED VIA CABIN ATTENDANT PANEL (CAP) (FULL SOLAR LOAD WITH DIMMED WINDOWS AND LIGHTING)
- LOWER RECIRCULATION FANS SELECTED OFF (UPPER FAN ON)
- ICS RECIRCULATION CHILLING OFF
- IFE OFF
- 100% OCCUPANT LOAD FACTOR

LEGEND:
- 440 SEATS
- 382 SEATS

NOTE:
THIS GRAPH PROVIDES THE FLOW RATE AND TEMPERATURE REQUIRED TO COOL A BULK AVERAGE TEMPERATURE OF 75 °F WHEN UTILIZING A PRE-CONDITIONED AIR PCA SOURCE(S).
5.6.13 Conditioned Air Flow Requirements – Cooling – Steady State (80 °F Ambient Air): Model 787-10

**CONDITIONS:**
- ALL EXTERIOR DOORS AND WINDOWS CLOSED
- "HEAT REDUCTION MODE" SELECTED VIA CABIN ATTENDANT PANEL (CAP) (FULL SOLAR LOAD WITH DIMMED WINDOWS AND LIGHTING)
- LOWER RECIRCULATION FANS SELECTED OFF (UPPER FAN ON)
- ICS RECIRCULATION CHILLING OFF
- IFE OFF
- 100% OCCUPANT LOAD FACTOR

**LEGEND:**
- 440 SEATS
- 382 SEATS

**NOTE:**
THIS GRAPH PROVIDES THE FLOW RATE AND TEMPERATURE REQUIRED TO COOL A BULK AVERAGE TEMPERATURE OF 75 °F WHEN UTILIZING A PRE-CONDITIONED AIR PCA SOURCE(S).
5.6.14 Conditioned Air Flow Requirements – Heating Time: Model 787-10

**CONDITIONS:**
- ALL EXTERIOR DOORS AND WINDOWS ARE CLOSED
- OUTSIDE AIR TEMPERATURE IS -40°F (-40°C)
- INITIAL CABIN TEMPERATURE IS -25°F (-32°C)
- NO SOLAR LOAD
- LOWER RECIRCULATION FANS SELECTED OFF (UPPER FAN ON)
- ICS RECIRCULATION CHILLING OFF
- NO OCCUPANTS
- IFE OFF
- NO ELECTRICAL HEAT LOADS

**LEGEND:**
- PCA TEMPERATURES AT GROUND CONNECTION
  - 120°F (48.9°C)
  - 140°F (60.0°C)
  - 160°F (71.1°C)

**NOTE:**
THIS GRAPH PROVIDES THE PREDICTED TIME REQUIRED TO HEAT THE AIRPLANE’S CABIN TO A BULK AVERAGE OF 75°F (24°C) AS A FUNCTION OF AIRFLOW AND TEMPERATURE, AT THE GROUND AIR CONNECTION, WHEN USING A PRE-CONDITIONED AIR (PCA) SOURCE.
5.6.15 Conditioned Air Flow Requirements – Heating – Steady State: Model 787-10

**CONDITIONS:**
- All exterior doors and windows are closed
- No solar heat load
- Lower recirculation fans selected off (upper fan on)
- ICS recirculation chilling off
- No occupants
- IFE off
- No electrical heat loads

**LEGEND:**
- 40°F (-40°C) AMBIENT
- 0°F (-17.8°C) AMBIENT

**NOTE:**
This graph provides the flow rate and temperature, at the ground air connection, that is required to heat the airplane’s cabin to a bulk average of 75°F (24°C) when using a pre-conditioned air (PCA) source.
5.7 GROUND TOWING REQUIREMENTS

5.7.1 Ground Towing Requirements - English Units: Model 787-8, 787-9, 787-10
5.7.2 Ground Towing Requirements - Metric Units: Model 787-8, 787-9, 787-10

NOTES:
1. Example: Shows a 787 weighing 220,445 kilograms being pushed up a 25% slope on sanded ice at 
-18°C backing against one engine at idle thrust. 1741 kilogram excess traction is required for towing.
2. Unusual breakaway conditions not shown.
4. Coefficients of friction (µ) are estimated for rubber-tired tow vehicles.

AIRPLANE GROSS WEIGHT - KILOGRAMS X 1,000

ENGINE RESISTANCE
RESISTANCE (NUMBER OF ENGINES BACKING AGAINST IDLE THRUST)

PERCENT SLOPE (%)

TOTAL TRACTION WHEEL LOAD - KILOGRAMS X 1,000

DRAWBAR PULL/PUSH - KILOGRAMS X 1,000
6.0 JET ENGINE WAKE AND NOISE DATA

6.1 JET ENGINE EXHAUST VELOCITIES AND TEMPERATURE

This section shows exhaust velocity and temperature contours aft of the 787 airplane. The contours were calculated from a standard computer analysis using three-dimensional viscous flow equations with mixing of primary, fan, and free-stream flow. The presence of the ground plane is included in the calculations as well as engine tilt and toe-in. Mixing of flows from the engines is also calculated. The analysis does not include thermal buoyancy effects which tend to elevate the jet wake above the ground plane. The buoyancy effects are considered to be small relative to the exhaust velocity and therefore are not included.

The graphs show jet wake velocity and temperature contours for representative engines. The results are valid for sea level, static, standard day conditions. The effect of wind on jet wakes is not included. There is evidence to show that a downwind or an upwind component does not simply add or subtract from the jet wake velocity, but rather carries the whole envelope in the direction of the wind. Crosswinds may carry the jet wake contour far to the side at large distances behind the airplane.

It should be understood, these exhaust velocity contours reflect steady-state, at maximum taxi weight, and not transient-state exhaust velocities. A steady-state is achieved with the aircraft in a fixed location, engine running at a given thrust level and measured when the contours stop expanding and stabilize in size, which could take several seconds. The steady-state condition, therefore, is conservative. Contours shown also do not account for performance variables such as ambient temperature or field elevation. For the terminal area environment, the transient-state is a more accurate representation of the actual exhaust contours when the aircraft is in motion and encountering static air with forward or turning movement, but it is very difficult to model on a consistent basis due to aircraft weight, weather conditions, the high degree of variability in terminal and apron configurations, and intensive numerical calculations. If the contours presented here are overly restrictive for terminal operations, The Boeing Company recommends conducting an analysis of the actual exhaust contours experienced by the using aircraft at the airport.
6.1.1 Jet Engine Exhaust Velocity Contours – Idle Thrust / Both Engines: Model 787-8
6.1.2 Jet Engine Exhaust Velocity Contours - Breakaway Thrust / 0%
Slope / Both Engines / MTW: Model 787-8
6.1.3 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 1% Slope / Both Engines / MTW: Model 787-8
6.1.4 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Single Engine / MTW: Model 787-8
6.1.5 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Single Engine / MLW: Model 787-8
6.1.6 Jet Engine Exhaust Velocity Contours – Takeoff Thrust / Both Engines: Model 787-8
6.1.7 Jet Engine Exhaust Temperature Contours – Idle/Breakaway Thrust: Model 787-8

Temperature contours for idle/breakaway power conditions are not shown as the maximum temperature aft of the 787-8 is predicated to be less than 100° F (38° C) for standard day conditions of 59° F (15° C).
6.1.8 Jet Engine Exhaust Temperature Contours – Takeoff Thrust / Both Engines: Model 787-8
6.1.9 Jet Engine Exhaust Velocity Contours – Idle Thrust: Model 787-9
6.1.10 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Both Engines / MTW: Model 787-9
6.1.11 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 1% Slope / Both Engines / MTW: Model 787-9

Temperature contours for idle/breakaway power conditions are not shown as the maximum temperature aft of the 787-9 is predicated to be less than 100° F (38° C) for standard day conditions of 59° F (15° C).
6.1.16  Jet Engine Exhaust Temperature Contours – Takeoff Thrust / Both Engines: Model 787-9
6.1.17 Jet Engine Exhaust Velocity Contours – Idle Thrust / Both Engines: Model 787-10
6.1.18  Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Both Engines: Model 787-10
6.1.19 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 1% Slope / Both Engines / MTW: Model 787-10
6.1.20 Jet Engine Exhaust Velocity Contours – Breakaway Thrust / 0% Slope / Single Engine / MTW: Model 787-10

**NOTES:**
- Engine thrust at breakaway setting
- Contours calculated from computer data
- Standard day
- Sea level
- No wind
- No slope
- Single engine running
- 7,540 lb (3,417 kg) engine (MTW=531,500 lb)

**Ground Plane:**
- 50 MPH (80 KM/H)
- 35 MPH (56 KM/H)

**Axial distance from aft of airplane:**
- 0, 25, 50, 75, 100, 125

**Distance from airplane centerline:**
- 0, 20, 40, 60, 80, 100
6.1.22 Jet Engine Exhaust Velocity Contours – Takeoff Thrust / Both Engines: Model 787-10
6.1.23 Jet Engine Exhaust Temperature Contours – Idle/Breakaway Thrust: Model 787-10

Temperature contours for idle/breakaway power conditions are not shown as the maximum temperature aft of the 787-10 is predicated to be less than 100° F (38° C) for standard day conditions of 59° F (15° C).
6.1.24 Jet Engine Exhaust Temperature Contours – Takeoff Thrust / Both Engines: Model 787-10
6.1.25 Inlet Hazard Areas: All Models

General Electric (GE) Engines

<table>
<thead>
<tr>
<th></th>
<th>RADIUS</th>
<th>AFT DISTANCE</th>
</tr>
</thead>
<tbody>
<tr>
<td>IDLE THRUST</td>
<td>15.0 FT</td>
<td>4.6 M</td>
</tr>
<tr>
<td>BREAKAWAY THRUST</td>
<td>28.5 FT</td>
<td>8.7 M</td>
</tr>
<tr>
<td>TAKEOFF THRUST</td>
<td>63.0 FT</td>
<td>19.2 M</td>
</tr>
</tbody>
</table>

Rolls Royce Engines

<table>
<thead>
<tr>
<th></th>
<th>RADIUS</th>
<th>AFT DISTANCE</th>
</tr>
</thead>
<tbody>
<tr>
<td>IDLE THRUST</td>
<td>15.0 FT</td>
<td>4.6 M</td>
</tr>
<tr>
<td>BREAKAWAY THRUST</td>
<td>28.5 FT</td>
<td>8.7 M</td>
</tr>
<tr>
<td>TAKEOFF THRUST</td>
<td>63.0 FT</td>
<td>19.2 M</td>
</tr>
</tbody>
</table>
6.2 AIRPORT AND COMMUNITY NOISE

Airport noise is of major concern to the airport and community planner. The airport is a major element in the community's transportation system and, as such, is vital to its growth. However, the airport must also be a good neighbor, and this can be accomplished only with proper planning. Since aircraft noise extends beyond the boundaries of the airport, it is vital to consider the impact on surrounding communities. Many means have been devised to provide the planner with a tool to estimate the impact of airport operations. Too often they oversimplify noise to the point where the results become erroneous. Noise is not a simple subject; therefore, there are no simple answers.

The cumulative noise contour is an effective tool. However, care must be exercised to ensure that the contours, used correctly, estimate the noise resulting from aircraft operations conducted at an airport.

The size and shape of the single-event contours, which are inputs into the cumulative noise contours, are dependent upon numerous factors. They include the following:

1. Operational Factors
   a. **Aircraft Weight** - Aircraft weight is dependent on distance to be traveled, en route winds, payload, and anticipated aircraft delay upon reaching the destination.
   b. **Engine Power Settings** - The rates of ascent and descent and the noise levels emitted at the source are influenced by the power setting used.
   c. **Airport Altitude** - Higher airport altitude will affect engine performance and thus can influence noise.

2. Atmospheric Conditions-Sound Propagation
   a. **Wind** - With stronger headwinds, the aircraft can take off and climb more rapidly relative to the ground. Also, winds can influence the distribution of noise in surrounding communities.
   b. **Temperature and Relative Humidity** - The absorption of noise in the atmosphere along the transmission path between the aircraft and the ground observer varies with both temperature and relative humidity.

3. Surface Condition-Shielding, Extra Ground Attenuation (EGA)
   a. **Terrain** - If the ground slopes down after takeoff or up before landing, noise will be reduced since the aircraft will be at a higher altitude above ground. Additionally, hills, shrubs, trees, and large buildings can act as sound buffers.

All these factors can alter the shape and size of the contours appreciably. To demonstrate the effect of some of these factors, estimated noise level contours for two different
operating conditions are shown below. These contours reflect a given noise level upon a ground level plane at runway elevation.

Condition 1

<table>
<thead>
<tr>
<th>Landing</th>
<th>Takeoff</th>
</tr>
</thead>
<tbody>
<tr>
<td>Maximum Structural Landing Weight</td>
<td>Maximum Gross Takeoff Weight</td>
</tr>
<tr>
<td>10-knot Headwind</td>
<td>Zero Wind</td>
</tr>
<tr>
<td>3° Approach</td>
<td>84 °F</td>
</tr>
<tr>
<td>84 °F</td>
<td>Humidity 15%</td>
</tr>
<tr>
<td>Humidity 15%</td>
<td></td>
</tr>
</tbody>
</table>

Condition 2

<table>
<thead>
<tr>
<th>Landing</th>
<th>Takeoff</th>
</tr>
</thead>
<tbody>
<tr>
<td>85% of Maximum Structural Landing Weight</td>
<td>80% of Maximum Gross Takeoff Weight</td>
</tr>
<tr>
<td>10-knot Headwind</td>
<td>10-knot Headwind</td>
</tr>
<tr>
<td>3° Approach</td>
<td>59 °F</td>
</tr>
<tr>
<td>59 °F</td>
<td>Humidity 70%</td>
</tr>
<tr>
<td>Humidity 70%</td>
<td></td>
</tr>
</tbody>
</table>

As indicated from these data, the contour size varies substantially with operating and atmospheric conditions. Most aircraft operations are, of course, conducted at less than maximum design weights because average flight distances are much shorter than maximum aircraft range capability and average load factors are less than 100%. Therefore, in developing cumulative contours for planning purposes, it is recommended that the airlines serving a particular city be contacted to provide operational information.

In addition, there are no universally accepted methods for developing aircraft noise contours or for relating the acceptability of specific zones to specific land uses. It is
therefore expected that noise contour data for particular aircraft and the impact assessment methodology will be changing. To ensure that the best currently available information of this type is used in any planning study, it is recommended that it be obtained directly from the Office of Environmental Quality in the Federal Aviation Administration in Washington, D.C.

It should be noted that the contours shown herein are only for illustrating the impact of operating and atmospheric conditions and do not represent the single-event contour of the family of aircraft described in this document. It is expected that the cumulative contours will be developed as required by planners using the data and methodology applicable to their specific study.
7.0 PAVEMENT DATA

7.1 GENERAL INFORMATION

A brief description of the pavement charts that follow will help in their use for airport planning. Each airplane configuration is depicted with a minimum range of five loads imposed on the main landing gear to aid in interpolation between the discrete values shown. All curves for any single chart represent data based on rated loads and tire pressures considered normal and acceptable by current aircraft tire manufacturer's standards. Tire pressures, where specifically designated on tables and charts, are at values obtained under loaded conditions as certificated for commercial use.

Section 7.2 presents basic data on the landing gear footprint configuration, maximum design taxi loads, and tire sizes and pressures.

Maximum pavement loads for certain critical conditions at the tire-to-ground interface are shown in Section 7.3, with the tires having equal loads on the struts.

Pavement requirements for commercial airplanes are customarily derived from the static analysis of loads imposed on the main landing gear struts. The charts in Section 7.4 are provided in order to determine these loads throughout the stability limits of the airplane at rest on the pavement. These main landing gear loads are used as the point of entry to the pavement design charts, interpolating load values where necessary.

The flexible pavement design curves based on the US Army Corp of Engineers Method and the rigid pavement curves based on the Portland Cement Association Design Method are no longer provided in Sections 7.5 and 7.7. Refer to the State's design standards for pavement design requirements. For US airports, refer to FAA Advisory Circular (AC) 150/5320-6, "Pavement Design" and pavement design program FAARFIELD for flexible and rigid pavement design requirements.

The Load Classification Number (LCN) curves are no longer provided in section 7.6 and 7.8 since the LCN system for reporting pavement strength is obsolete, being replaced by the ICAO recommended ACN/PCN system in 1983. For questions regarding the LCN system contact Boeing Airport Operations Engineering:

AirportCompatibility@boeing.com

The ACN/PCN system (Section 7.10) as referenced in ICAO Annex 14, "Aerodromes," 8th Edition, July 2018, provides a standardized international airplane/pavement rating system replacing the various S, T, TT, LCN, AUW, ISWL, etc., rating systems used throughout the world. ACN is the Aircraft Classification Number and PCN is the Pavement Classification Number. An aircraft having an ACN equal to or less than the PCN can operate on the pavement subject to any limitation on the tire pressure. Numerically, the ACN is two times the derived single-wheel load expressed in thousands of kilograms, where the derived single wheel load is defined as the load on a single tire inflated to 181 psi (1.25 MPa) that would have the same pavement requirements as the aircraft.
Computationally, the ACN/PCN system uses the PCA program PDILB for rigid pavements and S-77-1 for flexible pavements to calculate ACN values.

The ACR-PCR system (Section 7.11) follows the proposed amendment to ICAO Annex 14 replacing the current ACN-PCN system. ACR is the Aircraft Classification Rating and PCR is the Pavement Classification Rating. An aircraft having an ACR equal to or less than the PCR can operate on the pavement subject to any limitation on the tire pressure. Numerically, the ACR is two times the derived single-wheel load expressed in hundreds of kilograms, where the derived single wheel load is defined as the load on a single tire inflated to 218 psi (1.5 MPa) that would have the same pavement requirements as the aircraft.

The method of pavement evaluation is left up to the airport with the results of their evaluation presented as follows:

<table>
<thead>
<tr>
<th>PCR</th>
<th>PAVEMENT TYPE</th>
<th>SUBGRADE CATEGORY</th>
<th>TIRE PRESSURE CATEGORY</th>
<th>EVALUATION METHOD</th>
</tr>
</thead>
<tbody>
<tr>
<td>R = Rigid</td>
<td>A = High</td>
<td>W = No Limit</td>
<td>T = Technical</td>
<td></td>
</tr>
<tr>
<td>F = Flexible</td>
<td>B = Medium</td>
<td>X = To 254 psi (1.75 MPa)</td>
<td>U = Using Aircraft</td>
<td></td>
</tr>
<tr>
<td></td>
<td>C = Low</td>
<td>Y = To 181 psi (1.25 MPa)</td>
<td></td>
<td></td>
</tr>
<tr>
<td></td>
<td>D = Ultra Low</td>
<td>Z = To 73 psi (0.5 MPa)</td>
<td></td>
<td></td>
</tr>
</tbody>
</table>

ACN values for flexible pavements are calculated for the following four subgrade categories:

Code A - High Strength - CBR 15
Code B - Medium Strength - CBR 10
Code C - Low Strength - CBR 6
Code D - Ultra Low Strength - CBR 3

ACN values for rigid pavements are calculated for the following four subgrade categories:

Code A - High Strength, \( k = 552.6 \text{ pci} (150 \text{ MN/m}^3) \)
Code B - Medium Strength, \( k = 294.7 \text{ pci} (80 \text{ MN/m}^3) \)
Code C - Low Strength, \( k = 147.4 \text{ pci} (40 \text{ MN/m}^3) \)
Code D - Ultra Low Strength, \( k = 73.7 \text{ pci} (20 \text{ MN/m}^3) \)

ACR values at any mass on rigid and flexible pavements are calculated for the following four subgrade categories:
Code A - High Strength - Subgrade Modulus, $E = 200$ MPa (29,008 psi)

Code B - Medium Strength - Subgrade Modulus, $E = 120$ MPa (17,405 psi)

Code C - Low Strength - Subgrade Modulus, $E = 80$ MPa (11,603 psi)

Code D - Ultra Low Strength - Subgrade Modulus, $E = 50$ MPa (7,252 psi)
7.2 LANDING GEAR FOOTPRINT: MODEL 787-8, 787-9, 787-10

<table>
<thead>
<tr>
<th></th>
<th>UNITS</th>
<th>787-8</th>
<th>787-9</th>
<th>787-10</th>
</tr>
</thead>
<tbody>
<tr>
<td>MAXIMUM DESIGN</td>
<td>LB</td>
<td>503,500</td>
<td>563,000</td>
<td>561,500</td>
</tr>
<tr>
<td>TAXI WEIGHT</td>
<td>KG</td>
<td>228,383</td>
<td>255,372</td>
<td>254,692</td>
</tr>
<tr>
<td>PERCENT OF 50%</td>
<td>%</td>
<td></td>
<td>SEE SECTION 7.4</td>
<td></td>
</tr>
<tr>
<td>WEIGHT ON MAIN</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>GEAR</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>NOSE GEAR TIRE</td>
<td>IN.</td>
<td>40 x 16.0 R16 26PR</td>
<td></td>
<td></td>
</tr>
<tr>
<td>SIZE</td>
<td>PSI</td>
<td>187</td>
<td>182</td>
<td>182</td>
</tr>
<tr>
<td>NOSE GEAR TIRE</td>
<td>MPa</td>
<td>1.29</td>
<td>1.25</td>
<td>1.25</td>
</tr>
<tr>
<td>PRESSURE</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>MAIN GEAR TIRE</td>
<td>IN.</td>
<td>50 x 20.0 R22 34 PR</td>
<td>54 x 21.0 R23 38 PR</td>
<td></td>
</tr>
<tr>
<td>SIZE</td>
<td>PSI</td>
<td>228</td>
<td>229</td>
<td>224</td>
</tr>
<tr>
<td>MAIN GEAR TIRE</td>
<td>MPa</td>
<td>1.57</td>
<td>1.58</td>
<td>1.54</td>
</tr>
<tr>
<td>PRESSURE</td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
</tbody>
</table>
# 7.3 Maximum Pavement Loads: Model 787-8, 787-9, 787-10

\[ V_{\text{NG}} = \text{Maximum vertical nose gear ground load at most forward center of gravity} \]

\[ V_{\text{MG}} = \text{Maximum vertical main gear ground load at most aft center of gravity} \]

\[ H = \text{Maximum horizontal ground load from braking} \]

**Note:** All loads calculated using airplane maximum design taxi weight

---

## Table

<table>
<thead>
<tr>
<th>Airplane Model</th>
<th>Unit</th>
<th>Max Design Taxi Weight</th>
<th>( V_{\text{NG}} ) Static at Most Fwd C.G.</th>
<th>( V_{\text{NG}} ) Static + Braking 10 ft/sec² Decel</th>
<th>( V_{\text{MG}} ) Per Strut at Max Load at Static Aft C.G.</th>
<th>( H ) Per Strut at Instantaneous Braking (( \mu = 0.8 ))</th>
</tr>
</thead>
<tbody>
<tr>
<td>787-8</td>
<td>lb</td>
<td>503,500</td>
<td>54,716</td>
<td>85,086</td>
<td>229,798</td>
<td>78,194</td>
</tr>
<tr>
<td></td>
<td>kg</td>
<td>228,383</td>
<td>24,819</td>
<td>38,594</td>
<td>104,234</td>
<td>35,468</td>
</tr>
<tr>
<td>787-9</td>
<td>lb</td>
<td>563,000</td>
<td>51,369</td>
<td>80,601</td>
<td>259,574</td>
<td>87,434</td>
</tr>
<tr>
<td></td>
<td>kg</td>
<td>255,372</td>
<td>23,301</td>
<td>36,560</td>
<td>117,741</td>
<td>39,659</td>
</tr>
<tr>
<td>787-10</td>
<td>lb</td>
<td>561,500</td>
<td>42,193</td>
<td>68,209</td>
<td>261,787</td>
<td>87,201</td>
</tr>
<tr>
<td></td>
<td>kg</td>
<td>254,692</td>
<td>19,138</td>
<td>30,939</td>
<td>118,745</td>
<td>39,554</td>
</tr>
</tbody>
</table>
7.4 LANDING GEAR LOADING ON PAVEMENT

7.4.1 Landing Gear Loading on Pavement: Model 787-8
7.4.2 Landing Gear Loading on Pavement: Model 787-9

[Diagram showing landing gear loading on pavement for Model 787-9]
7.4.3 Landing Gear Loading on Pavement: Model 787-10
7.5 FLEXIBLE PAVEMENT REQUIREMENTS - FAA DESIGN METHOD

For the flexible pavement design refer to the FAA AC 150/5320-6 "Airport Pavement Design and Evaluation" and pavement design program FAARFIELD. Both are available on the FAA website:

FAA AC 150/5320-6F: https://www.faa.gov/airports/resources/advisory_circulars/
FAARFIELD: https://www.faa.gov/airports/engineering/design_software/
7.6 FLEXIBLE PAVEMENT REQUIREMENTS - LCN CONVERSION

The Load Classification Number (LCN) curves are no longer provided in section 7.6 and 7.8 since the LCN system for reporting pavement strength is obsolete, being replaced by the ICAO recommended ACN/PCN system in 1983. For questions regarding the LCN system contact Boeing Airport Compatibility Engineering:

AirportCompatibility@boeing.com
7.7 RIGID PAVEMENT REQUIREMENTS - PORTLAND CEMENT ASSOCIATION DESIGN METHOD

The rigid pavement requirements based on the Portland Cement Association method are no longer provided.
7.8 RIGID PAVEMENT REQUIREMENTS - LCN CONVERSION

The Load Classification Number (LCN) curves are no longer provided in section 7.6 and 7.8 since the LCN system for reporting pavement strength is obsolete, being replaced by the ICAO recommended ACN/PCN system in 1983. For questions regarding the LCN system contact Boeing Airport Compatibility Engineering:

AirportCompatibility@boeing.com
7.9 RIGID PAVEMENT REQUIREMENTS - FAA DESIGN METHOD

For the rigid pavement design refer to the FAA AC 150/5320-6 “Airport Pavement Design and Evaluation” and pavement design program FAARFIELD. Both are available on the FAA website:

FAA AC 150/5320-6: https://www.faa.gov/airports/resources/advisory_circulars/
FAARFIELD: https://www.faa.gov/airports/engineering/design_software/
7.10 ACN/PCN REPORTING SYSTEM - FLEXIBLE AND RIGID PAVEMENTS

To determine the ACN of an aircraft on flexible or rigid pavement, both the aircraft gross weight and the subgrade strength category must be known. On the chart in section 7.10.1, for a 787-8 aircraft with gross weight of 320,000 lb on a (Code A), the flexible pavement ACN is 34. Referring to 7.10.4, the same aircraft on a high strength subgrade rigid pavement has an ACN of 34.

The following table provides ACN data in tabular format similar to the one used by ICAO in the “Aerodrome Design Manual Part 3, Pavements.” If the ACN for an intermediate weight between maximum taxi weight and minimum weight of the aircraft is required, Figures 7.10.1 through 7.10.6 should be consulted. Linear interpolation of the ACN values between two weight points will provide an approximate ACN value.

<table>
<thead>
<tr>
<th>AIRCRAFT TYPE</th>
<th>MAXIMUM TAXI WEIGHT LB (KG)</th>
<th>MINIMUM WEIGHT *[1]</th>
<th>LOAD ON ONE MAIN GEAR LEG (%)</th>
<th>TIRE PRESSURE PSI (MPa)</th>
<th>ACN FOR RIGID PAVEMENT SUBGRADES – pci (MN/m²)</th>
<th>ACN FOR FLEXIBLE PAVEMENT SUBGRADES – CBR</th>
</tr>
</thead>
<tbody>
<tr>
<td>787-8</td>
<td>503,500 (228,383)</td>
<td>250,000 (113,398)</td>
<td>45.64</td>
<td>228 (1.57)</td>
<td>61 72 84 96 60 66 81 106</td>
<td>26 28 32 37 25 27 30 39</td>
</tr>
<tr>
<td>787-9</td>
<td>563,000 (255,373)</td>
<td>250,000 (113,398)</td>
<td>46.11</td>
<td>229 (1.58)</td>
<td>65 77 90 104 66 73 88 118</td>
<td>35 40 47 54 35 39 44 51</td>
</tr>
<tr>
<td>787-10</td>
<td>561,500 (254,692)</td>
<td>250,000 (113,398)</td>
<td>46.63</td>
<td>224 (1.54)</td>
<td>66 77 91 105 67 73 89 119</td>
<td>25 27 31 35 25 26 29 36</td>
</tr>
</tbody>
</table>

*[1] Minimum weight used solely as a baseline for ACN curve generation.
7.10.1 Aircraft Classification Number - Flexible Pavement: Model 787-8

NOTE: The minimum weight shown on the chart is solely for the ACN curve generation.
7.10.2 Aircraft Classification Number - Flexible Pavement: Model 787-9

NOTE: The minimum weight shown on the chart is solely for the ACN curve generation.
7.10.3 Aircraft Classification Number - Flexible Pavement: Model 787-10

NOTE: The minimum weight shown on the chart is solely for the ACN curve generation.
7.10.4 Aircraft Classification Number - Rigid Pavement: Model 787-8

**NOTE:** The minimum weight shown on the chart is solely for the ACN curve generation.
7.10.5 Aircraft Classification Number – Rigid Pavement: Model 787-9

NOTE: The minimum weight shown on the chart is solely for the ACN curve generation.
7.10.6 Aircraft Classification Number – Rigid Pavement: Model 787-10

NOTE: The minimum weight shown on the chart is solely for the ACN curve generation.
7.11 ACR/PCR REPORTING SYSTEM - FLEXIBLE AND RIGID PAVEMENTS

To determine the ACR of an aircraft on flexible or rigid pavement, both the aircraft gross weight and the subgrade strength category must be known. In the chart in Section 7.11.1, for a 787-8 aircraft with gross weight of 600,000 lb and high subgrade strength, the flexible pavement ACR is 697. In Section 7.11.4, for the same gross weight and subgrade strength, the rigid pavement ACR is 928.

The following table provides ACR data in tabular format. If the ACR for an intermediate weight between maximum taxi weight and the empty weight of the aircraft is required, Figures 7.11.1 through 7.11.6 should be consulted.

<table>
<thead>
<tr>
<th>AIRCRAFT TYPE</th>
<th>MAXIMUM TAXI WEIGHT (LB (KG))</th>
<th>LOAD ON ONE MAIN GEAR LEG (%)</th>
<th>TIRE PRESSURE (PSI (MPa))</th>
<th>ACR FOR RIGID PAVEMENT SUBGRADES</th>
<th>ACR FOR FLEXIBLE PAVEMENT SUBGRADES</th>
</tr>
</thead>
<tbody>
<tr>
<td>787-8</td>
<td>503,500 (228,383) 250,000 (113,398)</td>
<td>45.64 228 (1.57)</td>
<td>HIGH E = 200 MPa MEDIUM E = 120 MPa LOW E = 80 MPa ULTRA LOW E = 50 MPa</td>
<td>670 790 870 970 550 590 690 910</td>
<td>270 290 320 360 270 270 280 310</td>
</tr>
<tr>
<td>787-9</td>
<td>563,000 (255,373) 250,000 (113,398)</td>
<td>46.11 229 (1.58)</td>
<td>HIGH E = 200 MPa MEDIUM E = 120 MPa LOW E = 80 MPa ULTRA LOW E = 50 MPa</td>
<td>740 860 960 1070 620 660 750 970</td>
<td>260 280 310 340 270 270 280 290</td>
</tr>
<tr>
<td>787-10</td>
<td>561,500 (254,692) 250,000 (113,398)</td>
<td>46.63 224 (1.54)</td>
<td>HIGH E = 200 MPa MEDIUM E = 120 MPa LOW E = 80 MPa ULTRA LOW E = 50 MPa</td>
<td>740 870 970 1080 620 660 760 990</td>
<td>260 280 310 350 270 270 280 300</td>
</tr>
</tbody>
</table>

*Minimum weight used solely as a baseline for ACR curve generation.
7.11.1 Aircraft Classification Rating- Flexible Pavement: Model 787-8
7.11.2 Aircraft Classification Rating - Flexible Pavement: Model 787-9
7.11.3 Aircraft Classification Rating- Flexible Pavement: Model 787-10
7.11.4 Aircraft Classification Rating- Rigid Pavement: Model 787-8
7.11.5 Aircraft Classification Rating- Rigid Pavement: Model 787-9
7.11.6 Aircraft Classification Rating - Rigid Pavement: Model 787-10
8.0 FUTURE 787 DERIVATIVE AIRPLANES

Boeing's philosophy is to evaluate the derivative potential of its airplanes to provide capabilities that maximize value to our customers.

Decisions to design and manufacture future derivatives of an airplane depend on many considerations, including customer requirements. Along with many other parameters, airport facilities are considered during the development of any future airplane.
9.0 SCALED 787 DRAWINGS

The drawings in the following pages show airplane plan view drawings, drawn to approximate scale as noted. The drawings may not come out to exact scale when printed or copied from this document. Printing scale should be adjusted when attempting to reproduce these drawings. Three-view drawing files of the 787, along with other Boeing airplane models, can be downloaded from the following website:

http://www.boeing.com/airports
9.1 MODEL 787-8

9.1.1 Scaled Drawings – 1:500: Model 787-8

NOTE: WHEN PRINTING THIS DRAWING, MAKE SURE TO ADJUST FOR PROPER SCALING

LEGEND

A CONDITIONED AIR
B BULK CARGO DOOR
C CONTAINER CARGO DOOR
E ELECTRICAL
F FUEL
L LAVATORY
MG MAIN GEAR
NG NOSE GEAR
W POTABLE WATER
X PASSENGER DOOR
9.1.2 Scaled Drawings – 1:500: Model 787-8

NOTE: WHEN PRINTING THIS DRAWING, MAKE SURE TO ADJUST FOR PROPER SCALING
9.2 MODEL 787-9

9.2.1 Scaled Drawing – 1:500: Model 787-9

NOTE: WHEN PRINTING THIS DRAWING, MAKE SURE TO ADJUST FOR PROPER SCALING
9.2.2 Scaled Drawing – 1:500: Model 787-9

NOTE: WHEN PRINTING THIS DRAWING, MAKE SURE TO ADJUST FOR PROPER SCALING
9.3 MODEL 787-10

9.3.1 Scaled Drawings – 1:500: Model 787-10

NOTE: WHEN PRINTING THIS DRAWING, MAKE SURE TO ADJUST FOR PROPER SCALING
9.3.2 Scaled Drawings – 1:500: Model 787-10

NOTE: WHEN PRINTING THIS DRAWING, MAKE SURE TO ADJUST FOR PROPER SCALING